

Department of Physics and Astronomy

STATE UNIVERSITY OF IOWA

Iowa City, Iowa

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Final Engineering Report Mariner C - 64

Trapped Radiation Detector, Unit 25

JPL/University of Iowa contract 950613

when NAS 7-100

19 February 1965

Prepared by
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for

Jet Propulsion Laboratory 4800 Oak Grove Drive Pasadena, California

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1. Summary

Object of Report. The object of this report is to provide a concise technical description of the Mariner C program at the University of Iowa under JPL/University of Iowa Contract 950613, 17 May 1963 through 30 December 1964.

1.2 Scope of Work

- 1.2.1 Experiment Models. Design, fabricate, test, and deliver the following models of a Trapped Radiation Experiment:
 - a. 1 Engineering Prototype/Type Approval (EP/TA) Model.
 - b. 1 Proof Test Model (PTM/MC-1).
 - c. 3 Flight Models (MC-2, MC-3, MC-4).
 - d. 1 Set of Spare Parts (MC-5).
- 1.2.2 Bench Checkout Equipment. Design, fabricate, test, and deliver three sets of Bench Checkout Equipment (BCE) for the above experiment.

1.2.3 Services

- a. Establish engineering liaison with JPL.
- b. Furnish qualified personnel to perform engineering services at JPL and at ETR as required.
- 1.2.4 <u>Documentation</u>. Provide detailed technical and financial documentation for the project.
- Results. The U of I MC-2 experiment was launched aboard the MC-2 spacecraft (Mariner-3) on 5 November 1964. The mission was unsuccessful although the U of I experiment operated properly. MC-3 was launched aboard the MC-3 spacecraft (Mariner-4) on 28 November 1964. It has operated properly through 19 February 1965.

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5. General Description

- Mission Objectives. The University of Iowa Mariner C Trapped Radiation Experiment (Unit 25) is an array of particle detectors designed to search for and to measure any belts of trapped radiation possessed by Mars and to measure interplanetary cosmic rays while en route to Mars. These measurements will include flux, energy, directional characteristics, and identities of the particles.
- 5.2 General. In accordance with JPL/U of I contract 950613, the University of Iowa designed, constructed, tested, and delivered the following units and documentation:
 - of a subchassis containing dummy detectors and no electronics. The mass and size of the TCM were a close approximation of the subsequent flight units. The TCM was tested by JPL to determine the inherent thermal properties of unit 25 and to devise passive means of thermal control for the flight units.
 - Engineering Prototype/Type Approval (EP/TA).

 The EP/TA model was constructed as a fully operational unit except it possessed no means of thermal control. The instrument was subjected to, and passed the entire battery of environmental tests as called out in JPL EPD-59, specification number 30250 B.
 - Proof Test Model (MC-1). The PTM was the first flight quality instrument constructed. It was subjected to flight acceptance environmental tests (see EPD-59, Specification 30251 B). It was then used for integration and spacecraft testing.
 - 5.2.4 First Flight Unit (MC-2). This was the first flight unit. The unit was launched aboard the MC-2 spacecraft (Mariner 3) on 5 November 1964. The mission was unsuccessful although the SUI experiment apparently operated properly.
 - 5.2.5 Second Flight Unit (MC-3). MC-3, unit 25 was launched aboard the MC-3 spacecraft on 28 November 1964. It has operated properly through 19 February 1965.

- 5.2.6 Flight Spare MC-4. This instrument was intended to be used as a back up unit for flight units MC-2 and MC-3.
- 5.2.7 MC-5. This was an assemblage of spare subsystems and components of unit 25.
- Bench Checkout Equipment (BCE). Three complete sets were designed, constructed, assembled, and tested. The first two sets were delivered to JPL for bench testing of unit 25. The remaining unit was retained at the University of Iowa. Checkout sources were provided for EP/TA, MC-1, 2, 3, 4, 5.

5.2.9 Documentation

- a. Functional Specifications.
- b. Environmental Test Specifications.
- c. Design Specifications.
- d. Detail Drawings.
- e. Materials Lists.
- f. Flow Plans.
- g. Equipment Certification Data for Each Instrument.
- h. Quality Assurance Plan.
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- k. Failure Reports as Necessary.
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- m. Monthly Financial Reports.
- n. Final Engineering Report.
- 5.3 Weight. The total weight of this experiment: ~2.15 pounds.
- 5.4 Power. Power required: ~ 380 milliwatts (supplied as a ~ 2400 cps square wave).

- 5.5 Location on Spacecraft. The University of Iowa Mariner C Trapped Radiation Experiment was mounted on the Mariner C spacecraft in Section IV as is shown on JPL drawing number 4190502.
- 5.6 Configuration. The configuration of the experiment is a rectangular box 5" by 5.5" by 3". All of the particle detectors are mounted on one surface of this box and all of the subsystems contained inside (see University of Iowa drawing number 33-1003-C).

5.6.1 Detector View Angles

- a. Detector A: This detector has an unshielded window and a full view angle of 60° directed 135° from the probe-sun line. There are no obstructions in its field of view.
- b. Detector B: This detector has an unshielded window and a full view angle of 60°, directed 70° from the probe-sun line. There are no obstructions in its field of view.
- c. <u>Detector C</u>: This detector has a window shielded by approximately 20 mg/cm² of aluminum. View angle and direction are identical to Detector B.
- d. Detector D: View angle and direction are identical to Detector B.

6. Particle Detectors

- Geiger Mueller Counters. Detectors A, B, and C are type 6213 mica end window GM counters made by the Eon Corporation. Their energy threshold is nominally 40 keV for electrons and 500 keV for protons. The 6213 measures the total number of charged particles passing through its sensitive volume. The sensitive volume of each tube is shielded such that particles of low and moderate energies may only enter through the window. Much higher energy particles may enter from other directions. By allowing for the omnidirectional flux of high energy particles a directional measurement of the low energy particles is obtained.
 - Apertures. The diameter of the aperture of Detectors A, B, and C is 0.120 inches. Since the 6213 tube will count hard ultraviolet photons, each aperture is shaded from the sun by making the entrance corridor a 60° oblique truncated cone. The result is analogous to the visor on a cap (drawing number C35-0014C). Behind the aperture on Detector C is approximately 20 mg/cm² (~0.0025 in.) of aluminum. This serves to increase the threshold energy to 130 keV for electrons and 3 MeV for protons. Detectors A and B have no foil shields and are 40 keV/500 keV tubes.
- 6.2 PN Junction Detector (Silicon Surface-Barrier Diode).
 The solid state detector (Detector D) is a two-channel proton spectrometer. The detector and all of its associated electronics are enclosed in a gold plated RF tight box (University of Iowa drawing number D37-1001).
 - 6.2.1 Intended Objectives. Detector D measures flux and energy of protons and alpha particles within specific energy ranges while being essentially insensitive to electrons of all energies. It also makes possible the differentiation between the electrons and protons counted by the GM counters.

6.2.1.1 Functional Specifications. The full look angle of Detector D is 60° and the area is between 7-8 mm²; hence the geometric factor of the detector is about 0.06 cm² steradian.

The energy discrimination levels are: Channel D1, 500 keV and 11 MeV; Channel D2, 880 keV and 4 MeV.

Detector D contains an on-board Am 241 radioactive source of low intensity (< 0.01 μ C) to establish background counting rates. For Channel Dl, this rate is 3 to 4 counts per minute, and for Channel D2, it is 2 to 3 counts per minute.

The maximum counting rates of Channels Dl and D2 are about 35,000-41,000 counts per second.

- 6.2.2 Electron Sensitivity. The sensitivity to electrons of the detector is minimized (almost to zero) by three factors. First, the detector is too thin to absorb much energy from electrons. Second, the discrimination level is set much higher than the expected electron energy loss. Third, the pulses are clipped very short so as to avoid pile up of low energy electrons.
- 6.2.3 Detector Grounding Problem and Solution.

Early in the project, after the first few silicon surface barrier detectors were delivered to the University of Iowa, it became apparent under microscopic examination that the ground contact of the detector was not exceptionally good. The detectors on hand were selected on the basis of the appearance of a physically sound ground contact and the manufacturer was notified of the problem which he promised to correct.

However, the detector in MC-4, although apparently well grounded, failed at JPL. The failure was traced to the broken ground contact. This failure raised doubt as to the quality of the contacts on MC-2 and MC-3.

A conducting epoxy, Eccobond 56C (Emerson Cumming) was successfully used to adequately secure the ground connection on some sample detectors. The electrical characteristics of these detectors were equal to or better than they were originally. A few detectors did become noisy but seemed to "heal themselves" as time passed. Type-approval vibration tests were used in addition to a number of temperature cycles to test the permanence of the ground contact. There were no failures after being properly treated in this fashion.

The flight detectors for MC-2, MC-3, and MC-4 were all restored with this epoxy.

Although it was impossible to duplicate the intermittent ground and thus investigate the behavior of the detector, it appeared that the following is a possible sequence of events:

- (1) The ground contact was "half broken" in such a way that a very high resistance and capacitance appeared across the input of the charge sensitive preamplifier with the result that the amplifiers were driven into oscillation (output counting rates up to 40,000 c/s).
- (2) If the break in the ground contact was more or less "clean", i.e., open, the detector was not grounded at all. The pulses at the input of the preamplifier were greatly diminished so that the counting rate dropped to zero.

The above sequence of events fit well with the observed failure of MC-4. First a very high counting rate occurred, then a zero counting rate occurred. The diminished pulse-heights were observed at the output of the amplifiers.

Although the alleged failure of MC-2 was not duplicated on the bench, the data through the spacecraft system could be explained by reference to (2) above.

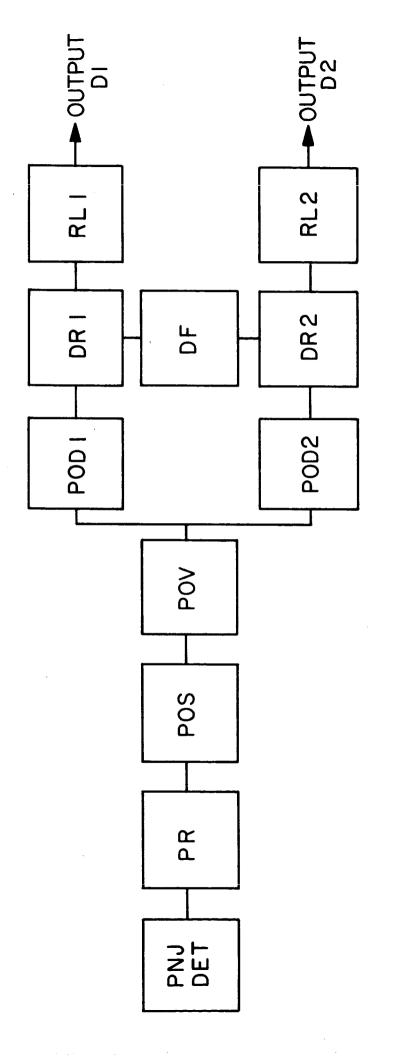
There were no further troubles of this nature.

7. Electronics

- 7.1 GM Preamplifiers. The output pulses from each of the GM counters is sent to one of the amplifiers. Each amplifier is a complementary NPN-PNP saturating device which shapes and conditions the GM pulses to meet the DAS interface requirements. These amplifiers were designed, fabricated, and tested by the University of Iowa (Drawing C33-3001).
- 7.2 PN Junction Detector Electronics. The overall operation of the detector, from data generation to data conditioning, may be understood by referring to Figure 1. All design, fabrication, and testing of these circuits was done by the University of Iowa (Drawing C37-2006).
 - 7.2.1 Preamplifier. A charged particle entering a solid state detector will release a number of electrons proportional to the energy of the particle which is lost in the detector. Taking this amount of charge into consideration along with the detector capacitance, a corresponding voltage pulse will occur at the output. The capacitance of a solid state detector varies when the detector pulse varies. However, charge released by the particle does not change with capacitance. This makes the charge sensitive amplifier a great deal more desirable than a voltage amplifier [Fillius, 1963] (Drawing C37-3005).
 - 7.2.2 Post Amplifiers. The post amplifiers of this instrument are highly stable negative feedback voltage amplifiers. They receive pulses from the preamplifiers, amplify them, and send them to the discriminators. The overall gain of each string of post amplifiers is adjusted so as to give voltage pulses at the discriminators of levels corresponding to the particles detected (Drawing 37-3007).
 - 7.2.3 <u>Discriminators</u>. These circuits are based on temperature stabilized voltage discriminators which accept voltage pulses above a pre-set level (> 1.4 volts). All pulses under this value are rejected (Drawing 37-3004).



FIGURE



PR = PREAMPLIFIER
POS = POSTAMPLIFIER
POV = POSTAMP (VARIABLE GAIN 4-20)
PODI = POSTAMP CHANNEL DI
POD2 = " " D2
DRI = DISCRIMINATOR CHANNEL DI
DR2 = " " D2
RL1 = RATE LIMITER DI
RL2 = " " D2
DF = DISCRIMINATOR REFERENCE PACK

25

7.2.4 Output and Rate Limiter. This circuit is based on a monostable multi-vibrator. It receives the data from the discriminators and serves to limit the maximum counting rates below 50,000 c/sec. Also incorporated are circuits which standardize the pulses to meet the Spacecraft Data Automation System (DAS) interface requirements (Drawing 37-3003).

8. Power Supply

The Unit 25 power supply was designed, fabricated, and tested by the Matrix Corporation of Nashua, New Hampshire to the University of Iowa and JPL specifications.

- 8.1 Configuration. The power supply was packaged such that all components were mounted on two, back-to-back, single sided printed circuit boards 2.75 inches by 3.50 inches. After the testing phase each power supply was potted, then foamed, which resulted in a vertical dimension of 1.50 inches. Weight before potting = 193 grams. Weight after potting = 282 grams. (Drawing C33-1002)
- 8.2 <u>Input</u>. The unit 25 power supply was driven by the space-craft power system as follows:

Frequency: 2400 cps square wave

Voltage: 50 v rms + 2%

Rise Time: 5 ± 4 usec

Experiment Power

Requirement: 400 milliwatts

- 8.2.1 <u>Input Limiter</u>. Incorporated in the design is a current limiter which limits the input current to below 400% of normal full load input current.
- 8.2.2 Fusing. The unit is also protected by a pair of 50 ma fuses, in parallel, inserted in the 50 VAC input. Since these fuses were installed late in the program they are not inside the power supply itself but on the back of the subchassis near the power connector (Drawing E33-1003-C).

8.3 Outputs

8.3.1 High Voltage Section. The high voltage section was designed to the following specifications:

Voltage: 900-1000 VDC unregulated

Current: 100 µa constant load

Ripple: 100 mV peak to peak maximum

This voltage served as the input to the corona

regulator tubes.

8.3.1.1 Corona Discharge Tubes. Corona regulators (VR tubes) were selected such that their operating voltage matched that of its associated GM counter. These regulator tubes and the entire high voltage section were potted in Eccosil silicone rubber prior to foaming the power supply with Eccofoam.

8.3.2 Detector D Bias Supply

Voltage: +75 VDC + 1%

Current: 50 µa

Ripple: < 1 mV high frequency

8.3.3 Circuit Supply

Voltage: + 8 VDC at 15 ma (set within

+ 0.2 V)

Current: 25 ma maximum

Ripple: < 2 mV

8.4 Corona Regulator Tube Pulsing Problem. Quite late in the project, certain anamolies were discovered in the data of MC-4 at low temperatures. The counting rates were high which is contrary to normal low temperature operation, i.e., low counting rates. It was discovered that the VR tubes were putting out pulses whenever their associated GM counter counted rapidly. A plausible explanation is this: Whenever the GM counter counts rapidly, the VR tube sees a burst of current pulses demanded by the GM tube driving the voltage below the regulating point. This triggers the VR tube into a regenerative region and the anode rapidly drops ($< 1 \mu sec$) to near zero volts before it can recover. The best solution found to this problem is to connect a capacitor across the VR tube. This capacitor serves to supply the current pulses required without an appreciable drop in voltage.

This corrective measure was taken in MC-4 but not in any of the other units.

9. Mechanical Design

9.1 Experiment Subchassis. The experiment subchassis was machined from cold-rolled magnesium alloy AZ-31B-H24. The box is divided into an upper and a lower compartment by a bulkhead to which are affixed most of the subsystems and components (Drawing E33-0001E). The subchassis for the TCM and the EP/TA were built by the University of Iowa. The subchassis for MC-1, 2, 3, 4, and 5 were built by the New Era Industries Company, Cedar Rapids, Iowa.

An error in the location of the mounting holes in the feet of the MC-1, 2, 3, and 4 subchassis was corrected by the following method: The holes which were improperly located were drilled and tapped to 5/16-24 NF thread. A threaded plug of magnesium was screwed into the holes. A small hole was then drilled lengthwise down through the threads and a pin inserted into this hole to secure the plug in the hole. The mounting holes were then redrilled in the correct location.

- 9.2 PN Junction Structure. The PN junction structure and covers were machined from cold rolled magnesium alloy ZA-31B-H24. The top and bottom covers are removable and are shielded against radio frequency interference by money-metal "rope" gaskets.
- Detector Housings, Structural Components, and Special Hard-ware. Detector housings and structural components were fabricated from cold rolled magnesium alloy AZ-31B-H24. The PN junction aperture was fabricated from non-magnetic brass. Special screws were fabricated from non-magnetic brass and non-magnetic stainless steel. See drawings for construction details. Standard screws were of titanium, furnished by JPL.
- 9.4 Surface Treatment. The subchassis for the TCM and EP/TA experiments were surface treated with Dow 7 dichromate at the University of Iowa. The subchassis for the MC-1, 2, 3, 4, and 5 models were treated with grit blast and dichromate per JPL specification 90433.

The PN junction structure was gold plated to provide good conduction between covers and structure and thus provide protection against RF interference.

Grit blasting, dichromating, and gold plating of MC-1, 2, 3, 4, and 5 components were done by Anchor Plating and Tinning Co., South El Monte, California.

GM tube housings on TCM and EP/TA were treated with Dow 7 dichromate and later gold plated. Those on MC-1, 2, 3, 4, and 5 were gold plated without the dichromate. Bores of the tube housings were treated with Dow 19.

9.4.1 Gold Plating. Early in the program it was discovered that pin holes were appearing in almost all gold plated magnesium surfaces.

Microscopic examination indicated that copper or silver sub-plating was "bleeding" through the gold exposing the under surface to the atmosphere. These surfaces were replated, still maintaining the 0.0005" minimum of copper and 0.0005" minimum of silver but increasing the gold to 0.0003".

(JPL specifications originally called for 0.0001".)

No further trouble of this nature occurred.

10. Thermal Design

General Information. Unit 25 was located on the upper ring of the spacecraft octagon section and slaved to its temperature. The spacecraft thermal shell was shaped such that the ends of the four detectors and the outboard face of the experiment were exposed to space. The outboard face of the chassis was painted with white paint (PV-100) to achieve the desired temperature range of approximately 100°F near earth and 50°F near Mars.

The GM tube housing and PN junction aperture on MC-1, 2, 3, and 4 were gold plated. Indium foil was used between the GM tube housings and the subchassis to improve thermal conductivity on MC-2, 3, and 4.

The bottom half of the PNJ aperture was painted white in order to lower the temperature of this detector to about 82° F near earth and 37° F near Mars.

Thermal design was performed by Jet Propulsion Laboratory.

10.2. Temperature Transducer. A temperature transducer (TS1 T4086S1) was provided by JPL and installed in each experiment at the University of Iowa. Identification of the transducers installed in various units are given in Table I.

Experiment	Table I	Transducer Serial Number
EP/TA MC-1 MC-2 MC-3 MC-4		41873 41872 41871 41870 41869

10.3 Temperature Control Model (TCM). A temperature control model was delivered to JPL for use in the thermal design of the spacecraft and the SUI experiment. The TCM was similar to the flight models in weight and configuration but contained no electronic subassemblies.

11. Bench Checkout Equipment (BCE)

Description. The bench checkout equipment (BCE) for the University of Iowa Mariner C experiment was especially simple since no timing signals were required to operate it. All detectors are operating whenever power is applied.

The BCE consists of a Glentronics, Inc., AC power supply to simulate the spacecraft power system, a Beckman electronic counter to count detector output pulses, a Tektronix oscilloscope to monitor the electrical properties of those pulses, a Beckman printer to record permanently the counter output, and a push button assembly to select the desired detector output. Only one detector was monitored at a time.

This equipment was mounted in a standard 19-inch rack (see II-20).

12. Quality Assurance and Control

Quality assurance and control was provided by adherence to appropriate U of I and JPL specifications listed in Section 14 of this report. Screened electronic components were provided by JPL. These components were used in MC-2, 3, 4, and 5. Components, unscreened by JPL, but tested in accordance with normal U of I practice were used in EP/TA and MC-1.

University of Iowa and JPL inspectors performed inprocess and final inspections on all modules, subassemblies and completed experiments.

- 12.1 <u>Visual Inspection and Temperature Tests</u>. Testing of the University of Iowa Mariner C experiment was done at various stages of fabrication as shown below:
 - a. Component: Visual and microscopic inspection of all components in accordance with applicable U of I and JPL specifications.
 - b. Module: Visual and microscopic inspection.

 Preliminary electrical tests over temperature range of -20° C to +65° C before and after conformal coating.
 - c. Subassembly: Visual and microscopic inspection.

 Electrical tests over temperature range of -20° C to + 65° C.
 - d. Experiment: Visual inspection. Electrical check
 --all detectors, all counting rates.
 Temperature test: -10° C to +50° C operating
 -30° C to +60° C survival.
- Vibration Testing. Vibration testing was performed at the University of Iowa, according to JPL Specifications 30250B and 30251B, on each experiment before being delivered to JPL.

In the event any module had to be replaced, a second vibration test was performed. This test was run to the same g-levels but of time duration one-half of that called for in the JPL Specification.

13. Experiment Information and Test Data

13.1 Weight and Power of Specific Experiments.

Listed in Table II is weight and power information concerning each experiment (measured at JPL by D. Schofield of JPL).

Experiment	Table II Weight (lbs.)	Power (watts)
EP/TA	2.10	0.36
MC-1	2.34	0.38
MC-2	2.13	0.38
MC-3	2.20	0.35
MC-4	2.15	0.36

13.2 Test Sources and Expected Counting Rates.

Sources and expected counting rates of the experiments with their assigned sources in standard source holders are given in Table III. The sources used were as follows:

<u>GM Checkout</u>---Co 60; ~ 200 μ C; 1.3 MeV γ

PNJ Checkout--Po210; \angle 10 $\not\vdash$ C; 5.5 MeV α , 0.6 MeV γ

Table III

Funoriment		ource	PNJ S	PNJ Source	Date of	France	2000			
חווסשד זהלעה	s/N	aς	S/N	277	Data Run	A	a counting	g kates	A B R Counting Rates (Counts/Second)	- 1
EP/TA*	c-62	223	Unmarked	< 10	July 64	6.1	74.7	21.7	80.4	80.3
MC-1*	89-2	206	٣	¥ 10	Aug. 64	9.8	156.5	43.9	**5#	***
MC-2	69-0	190	α	V 10	April 64	12.1+1.2	161+8.	64.4+1.	64.4+1.9 77.3+7	6.5
MC-3	0 2- 2	185	7	< 10	May 64	12.1+2.5	26.8+5**	*153.1+8*	26.8+5***153.1+8****50.045 6	11.01
MC-4	C-71	190	ω	× 10	June 64	9.5±1.5	145+7	74.843.	54.843.5 49.645.5	48.7+5.5
							}	ł	i	

EP/TA and MC-1 counting rates measured at JPL at times indicated. All other counting rates measured at the University of Iowa.

** The PNJ detector was replaced in April 1964 with one of smaller area. The higher counting rates occurred prior to April 1964.

Detectors B and C were physically interchanged on MC-3. They are electrically identical to
all other units. The source normally is placed on the "middle" GM tube and thus the difference

14. List of Applicable Documents

14.1 University of Iowa Documents

- a. Proposed Experiment for Mariner C 1964 Missions March 4, 1963.
- b. Mariner C-64 Functional Specifications for Trapped Radiation Detector--Unit 25 20 June 1963 including Amendment I of 12 September 1963.
- c. Design Specification Mariner C-64 Trapped Radiation Detector, Unit 25 30 August 1963 including Amendment 1 of 12 August 1964.
- d. Test Specification Mariner C-64 Trapped Radiation
 Detector, Unit 25
 12 August 1963.
 Amendment to this specification of 24 December 1963
 was later rescinded.
- e. Mariner Mars 64 Parts List 24 February 1964 Including Mariner C-64 Power Supply Parts List (Matrix Corporation).
- f. Mariner C-64 List of Drawings Rev. C3 September 1963.
- g. Quality Control Requirements for Mariner C-64
 Experiment
 26 September 1963.

14.2 <u>Jet Propulsion Laboratory Documents</u>

- a. Procedures and Requirements for Scientific Instruments, Mariner Mars-64, EPD-59.
- b. JPL Spec. 30250B, Environmental Specification, Mariner C Flight Equipment, Type Approval Environmental Test Procedures (Assembly Level).
- c. JPL Spec. 30251B, Environmental Specification, Mariner C Flight Equipment, Flight Acceptance Environmental Test Procedures and Preacceptance Test Limits (Assembly Level).

15. <u>Project Personnel</u>

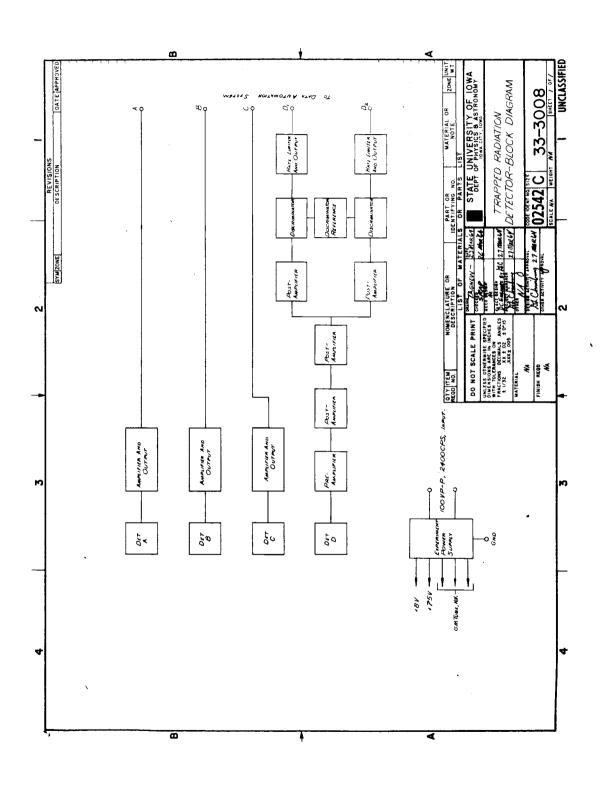
University of Iowa personnel directly involved in the conception, planning, design, fabrication, testing, and delivery of this experiment are as follows:

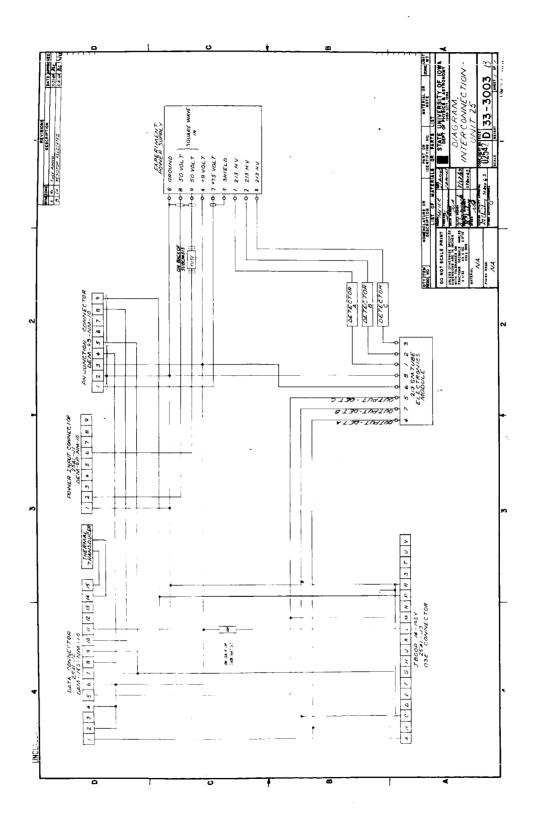
J. A. Van Allen	Principal Investigator	General director of the project.
T. Armstrong	Experimenter	PNJ detectors: testing, calibration, and field support.
D. L. Chinburg	Project Manager	Responsible for overall conduct of the project.
J. D. Craven	NASA Trainee	GM detectors: testing, calibration, and delivery.
G. Crossett	Designer	Packaging designs and layouts of electronic circuitry.
D. C. Enemark	Senior Design Engineer	Responsible for design of all detector electronics. Supervised all electronics, fabrication, and testing.
L. A. Frank	Co-Investigator	Responsible for scientific design criteria for GM detectors; procurement, testing, calibrating, and delivery of GM detectors.
E. A. Freund	Instrument Shop Supervisor	Responsible for all local fabrication of mechanical parts.
R. H. Gabel	Project Coordinator	Performed detailed coordination of design, fabrication, testing, delivery, and field support of the Mariner experiment.
R. A. Ganfield	Technician	PNJ testing and calibration, BCE fabrication.
J. S. Hathaway	Technician	Responsible for the quality control of the project.

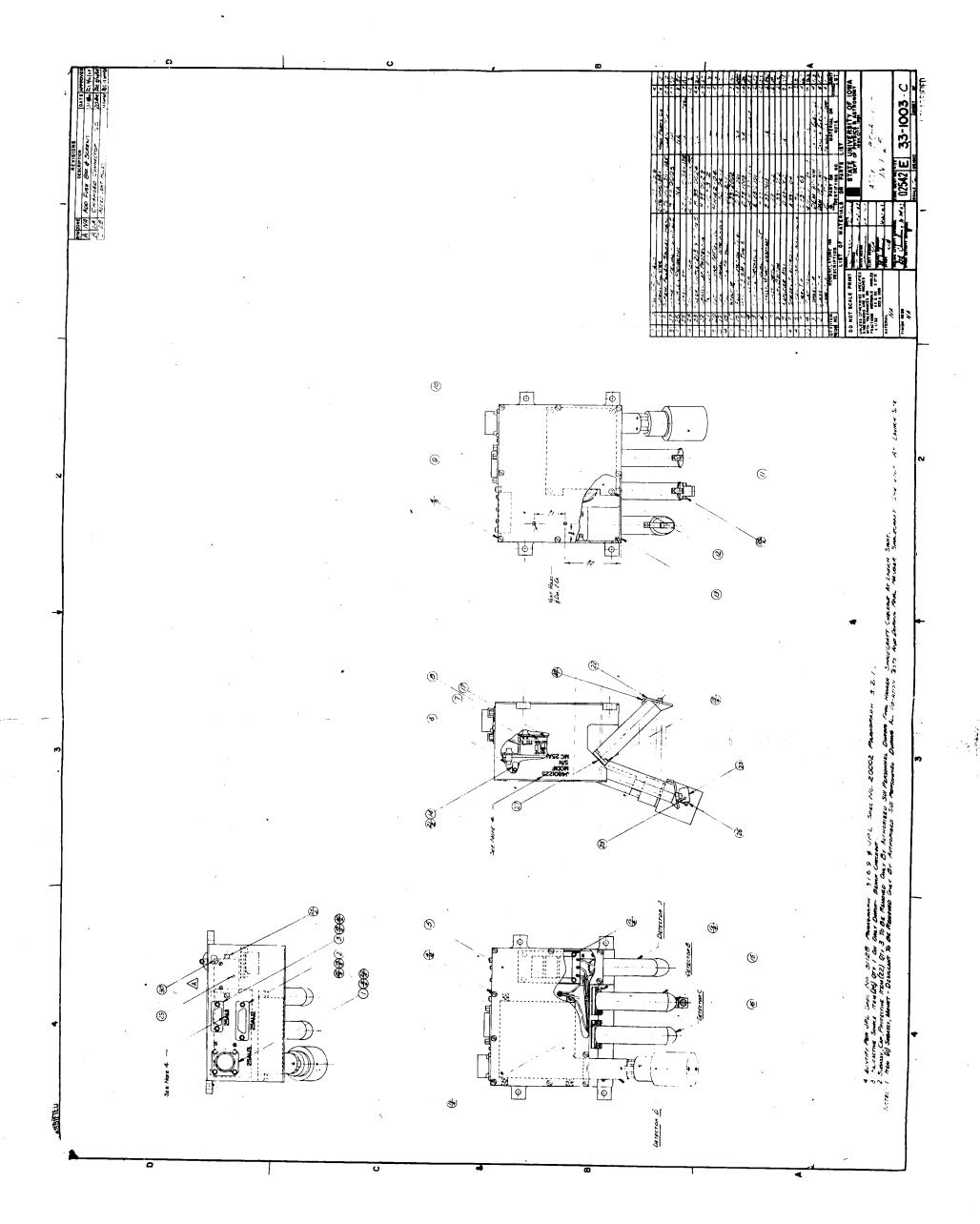
N.	K. Henderson	Techn icia n	Testing, calibration, delivery, and field support of GM detectors.
н.	K. Hills	NASA Trainee	GM detector and associated voltage regulator tube procurement, testing, calibration, delivery, and field support.
F.	Kolpin	Technician	Assistant to the Project Manager.
s.	Krimigis	Co-Investigator	Responsible for scientific design criteria for PN junction detector and associated electronics. Procurement, testing, calibrating, and field support of PNJ detectors.
E.	Lawrence	Mechanical Designer	Responsible for design, fabrication, and finishing of mechanical parts, and preparation of all finished drawings.
м.	Murphy 1	Electronics Shop Supervisor	Responsible for assembly of all electronics modules and harnessing of the experiments.
s.	R. A. Robinson	Mechanical Engineer	Responsible for general design, vibration analysis, and testing of the experiment.
м.	Schug	Technician	Assistant to the Project Co- ordinator.
T.	Schweitzer	Technician	Project quality control.
J.	F. Skerik	Project Administrator	Assistant to Project Manager.
W.	Stanley	Technici a n	Project quality control, testing, and field support.
E.	W. Strein	Technici a n	Construction and testing of PNJ detectors; packaging design of PNJ electronics; supervised design and construction of the BCE.
R.	Wenman	Electronics Shop Technician	Assembly of electronics modules and harnessing of the experiments.

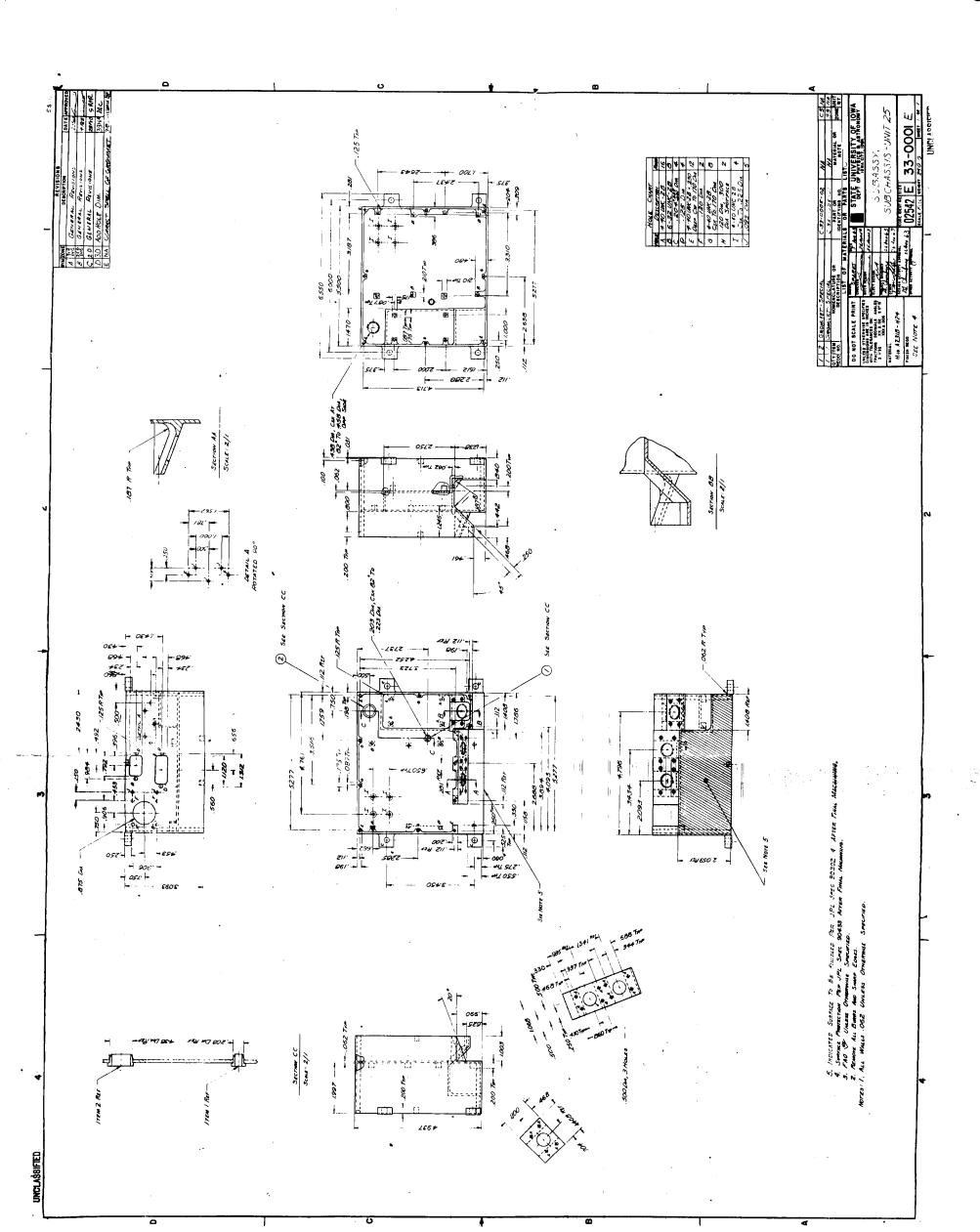
Appendix I

 $\underline{\mathtt{D}} \ \underline{\mathtt{R}} \ \underline{\mathtt{A}} \ \underline{\mathtt{W}} \ \underline{\mathtt{I}} \ \underline{\mathtt{N}} \ \underline{\mathtt{G}} \ \underline{\mathtt{S}}$

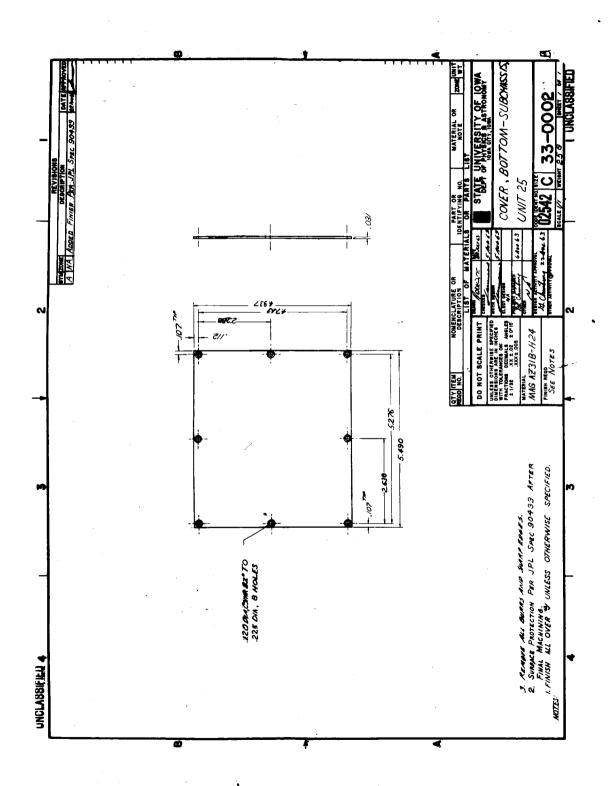


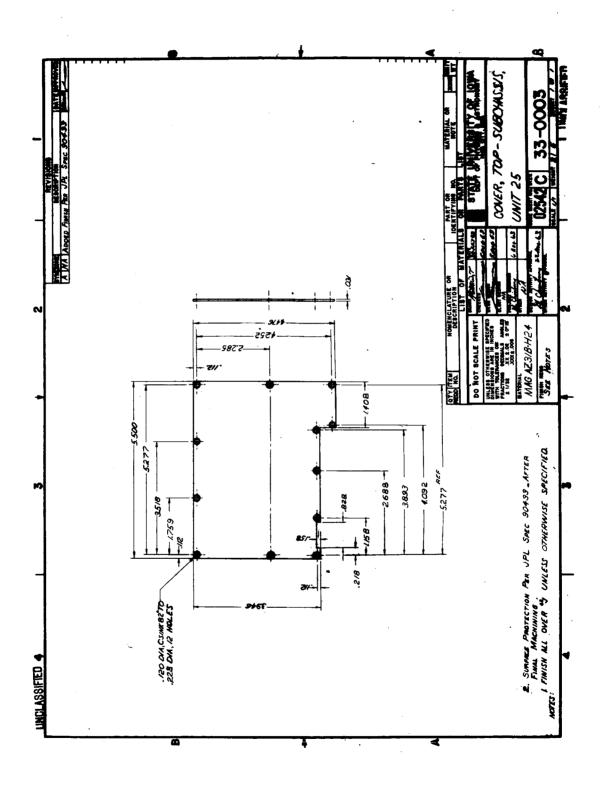


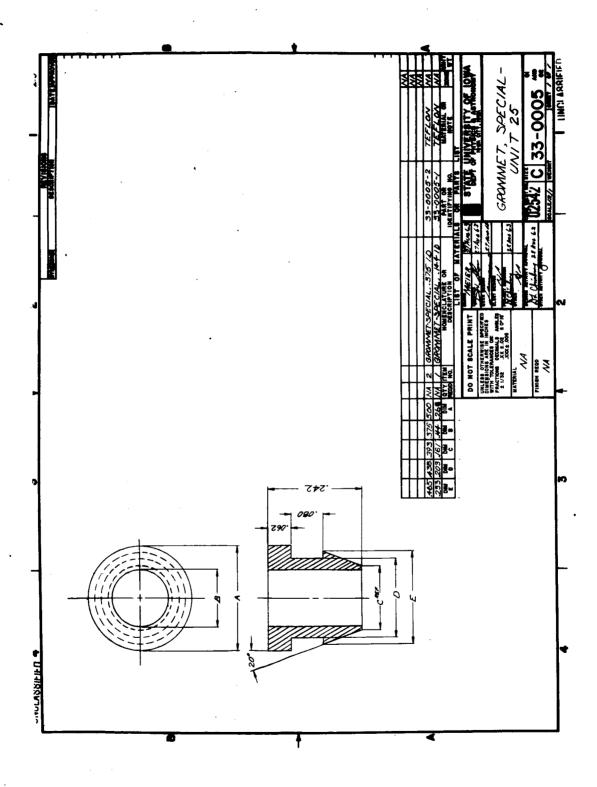


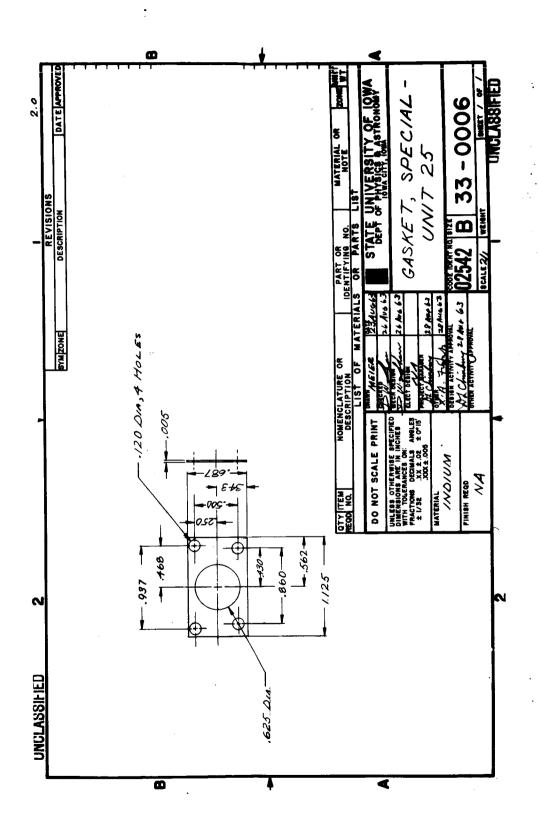


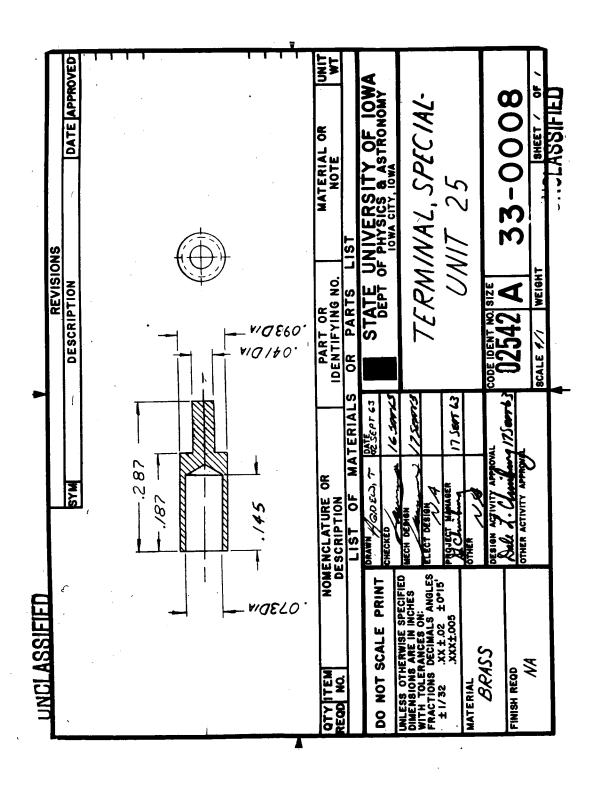
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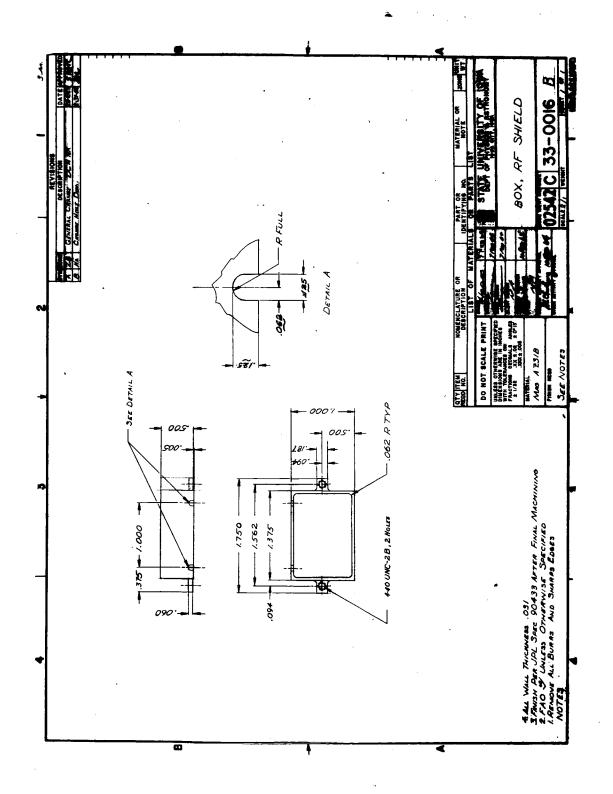


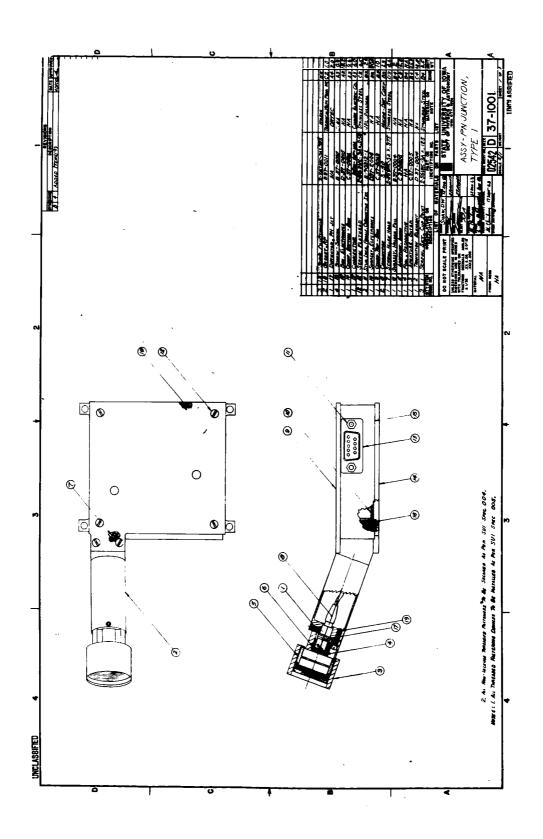


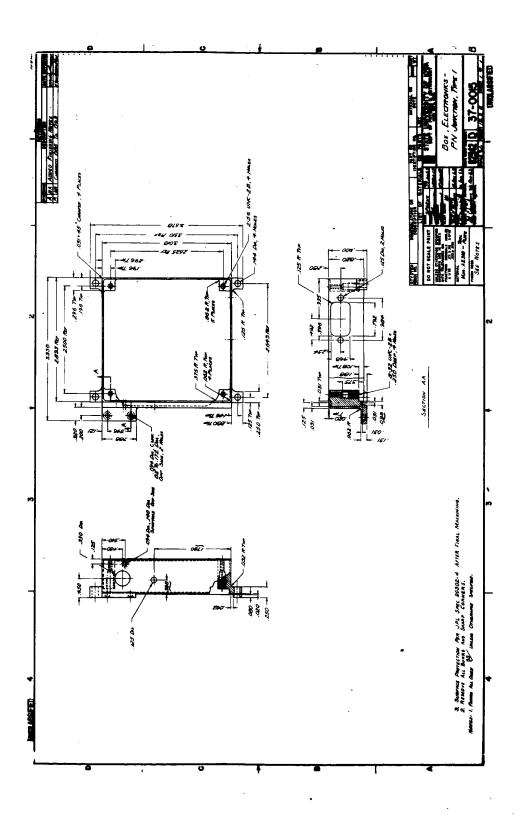


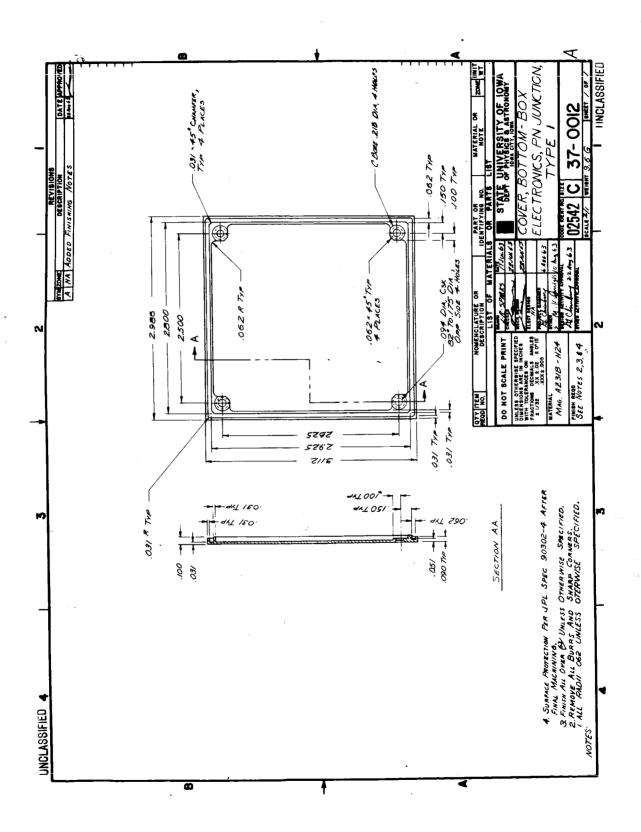


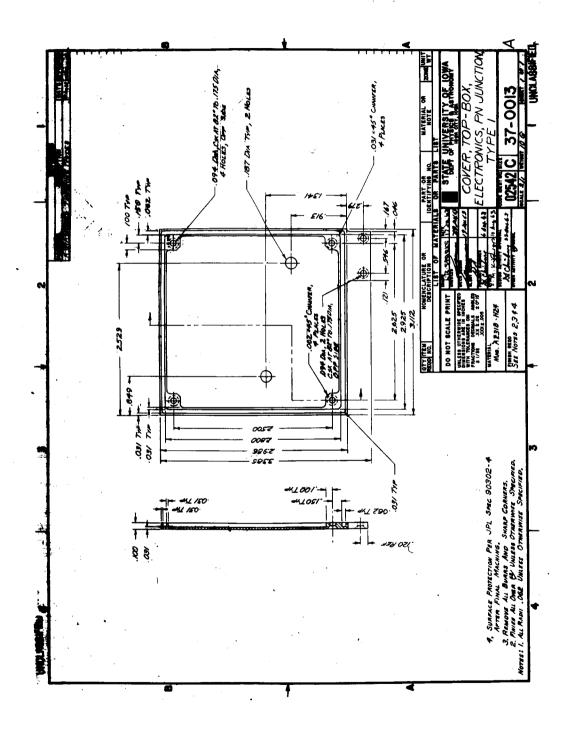


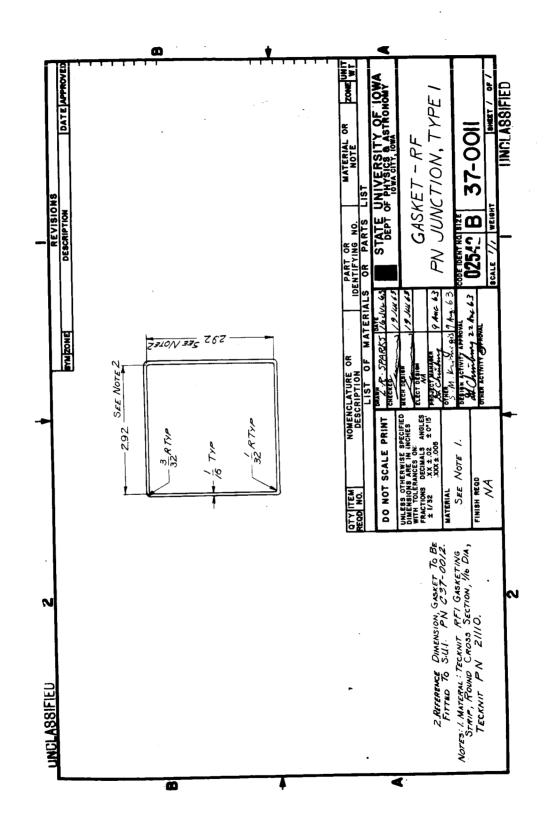


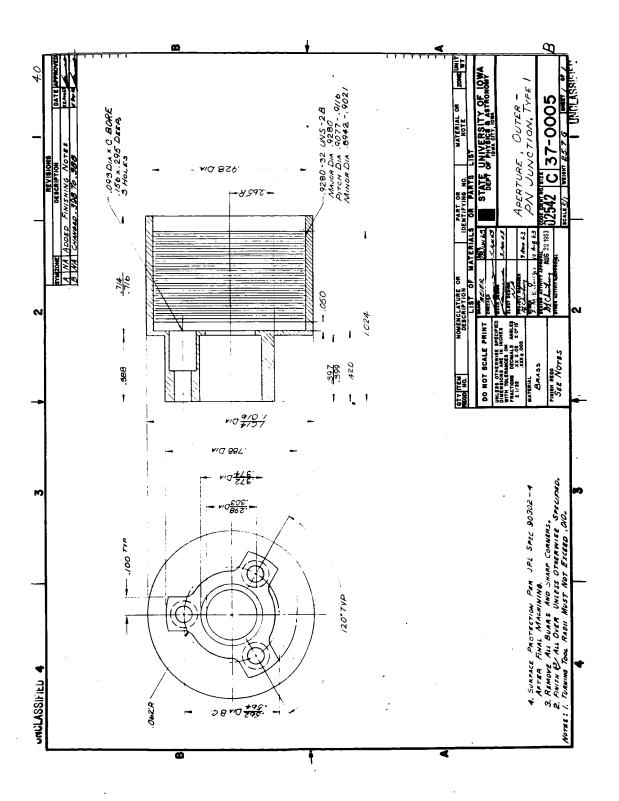


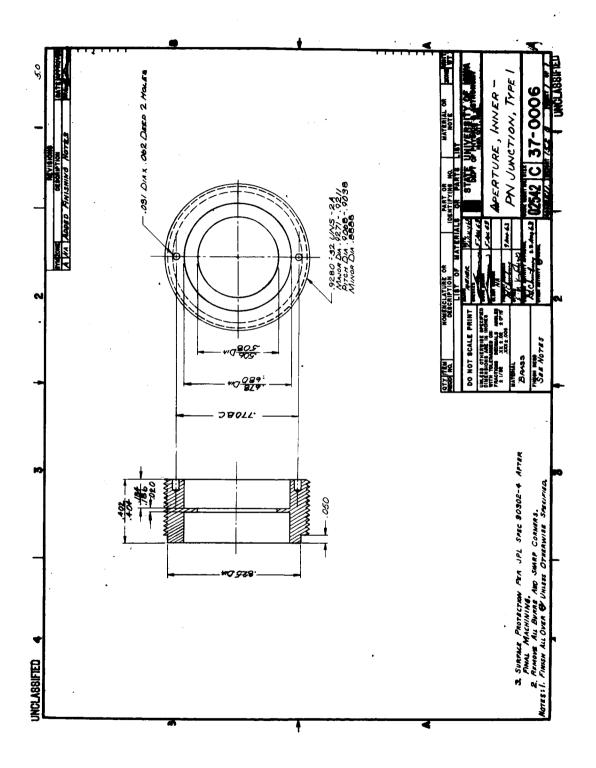


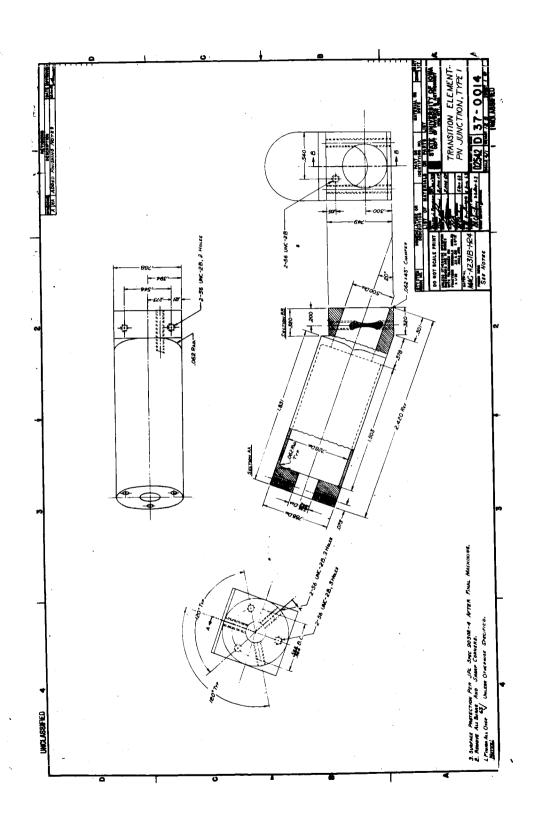


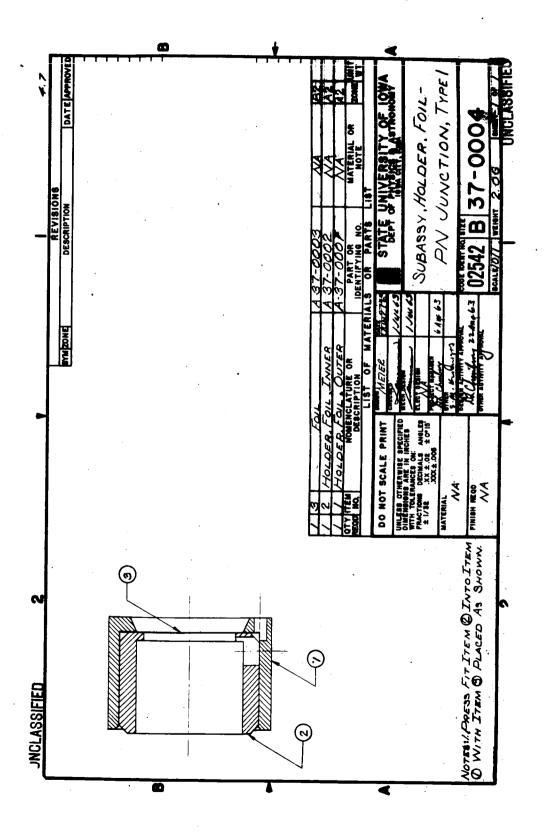


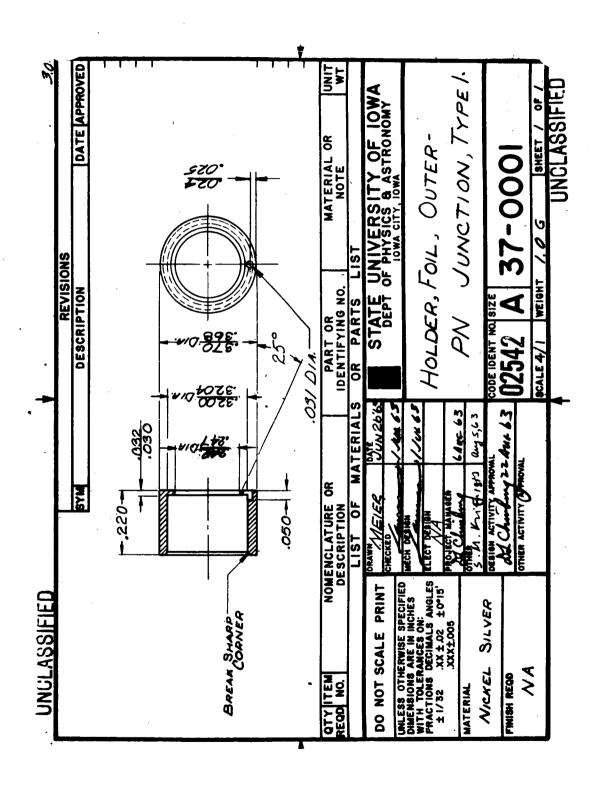


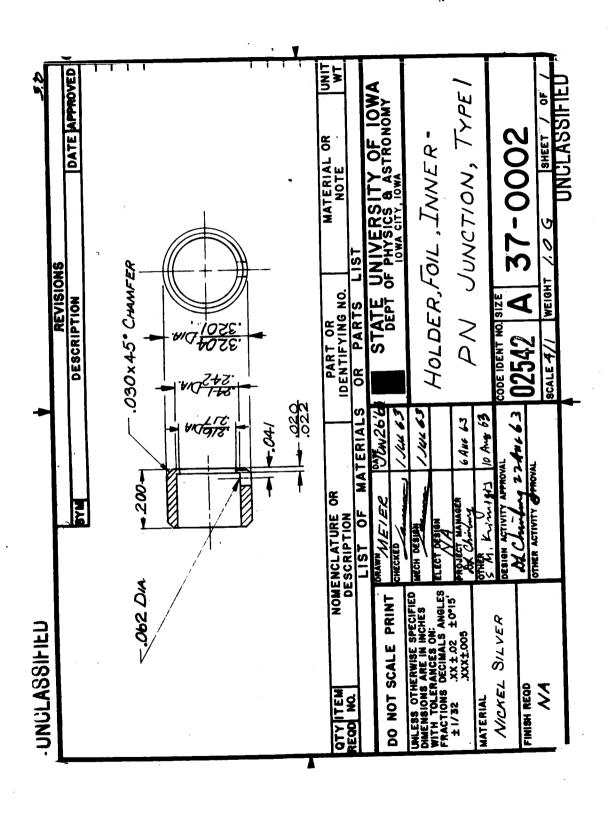




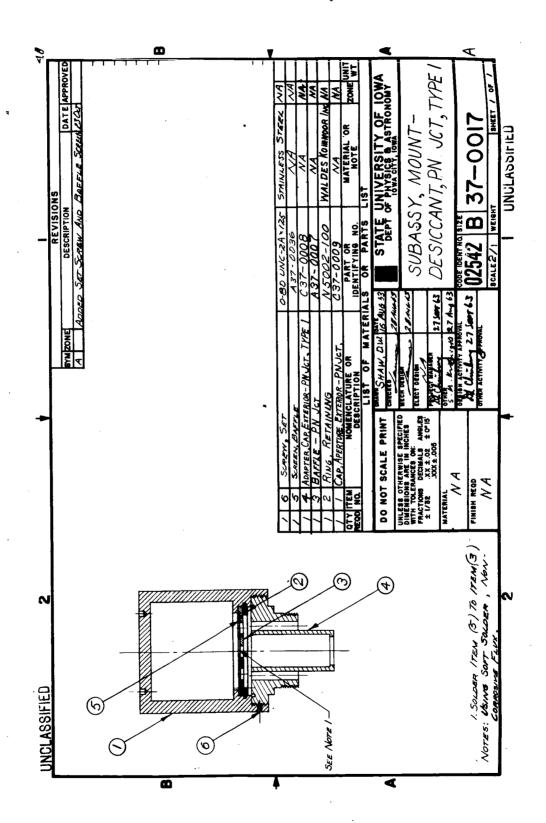


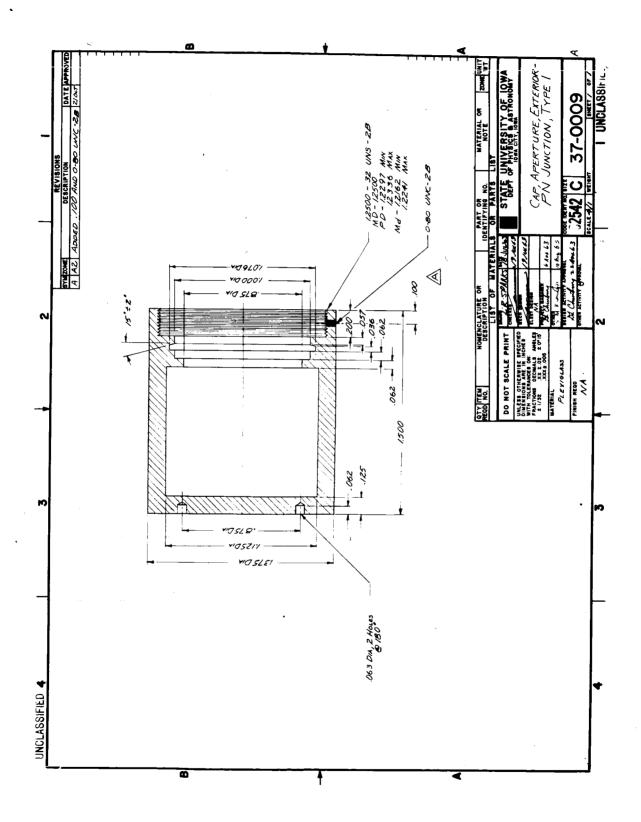


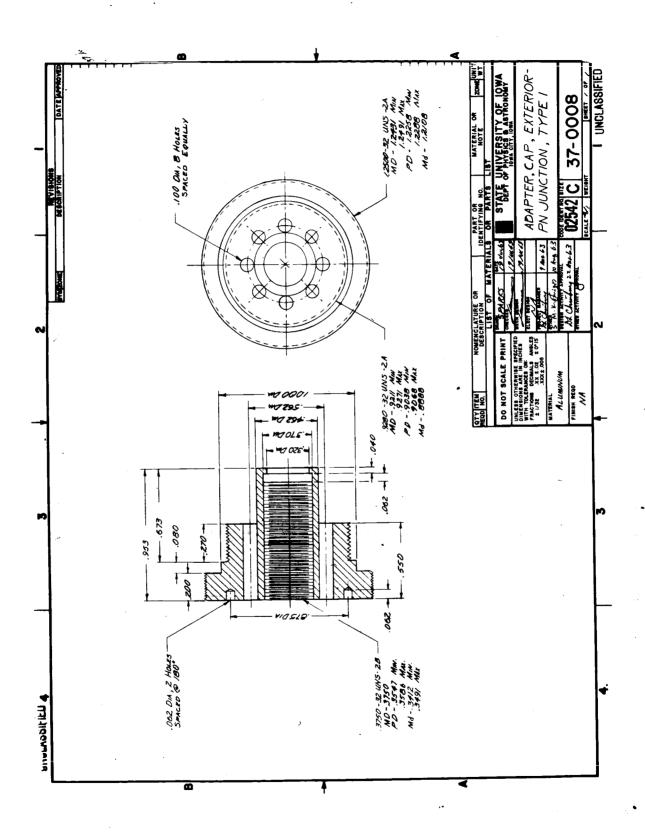


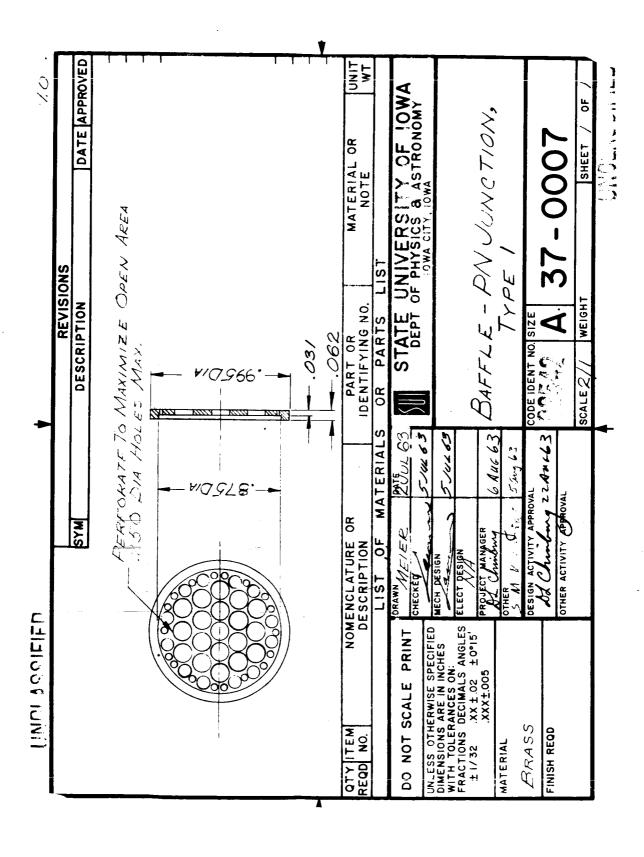


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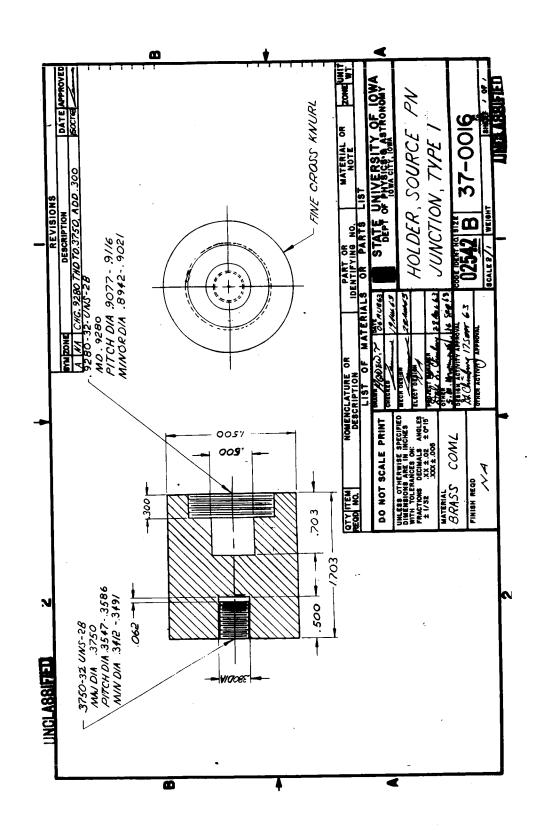


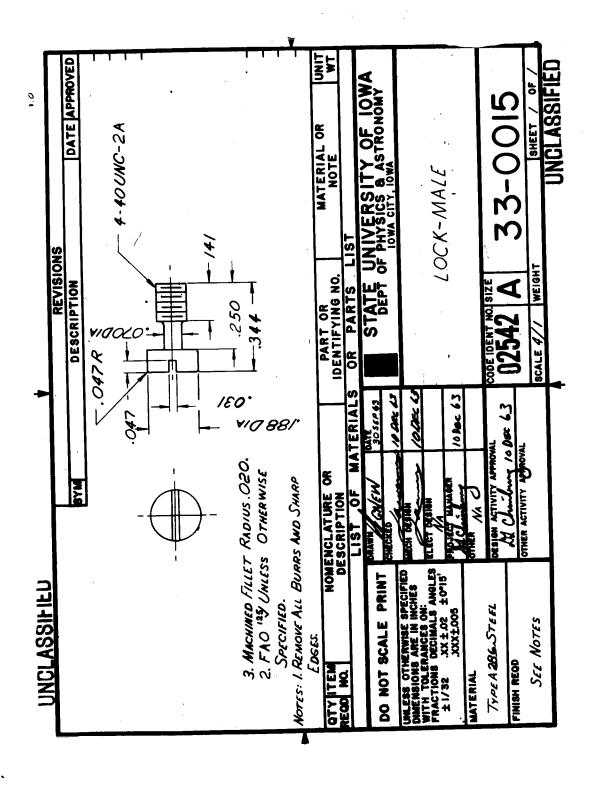


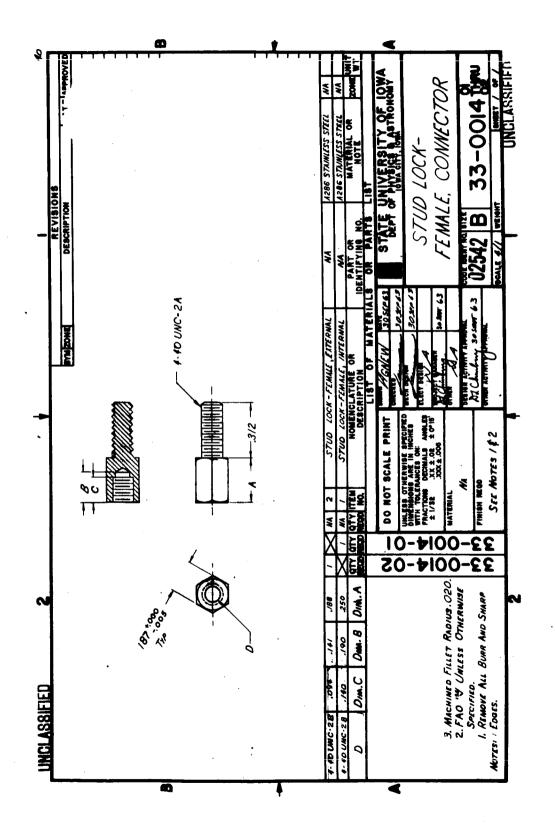


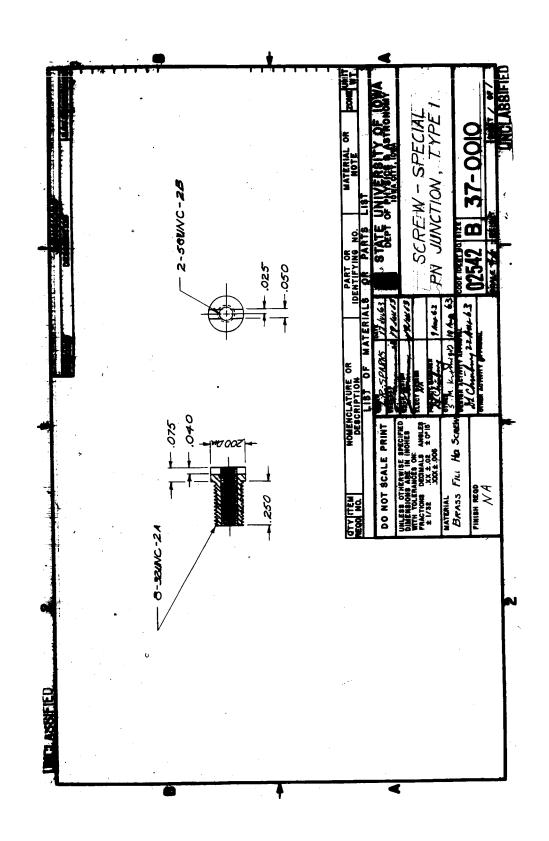


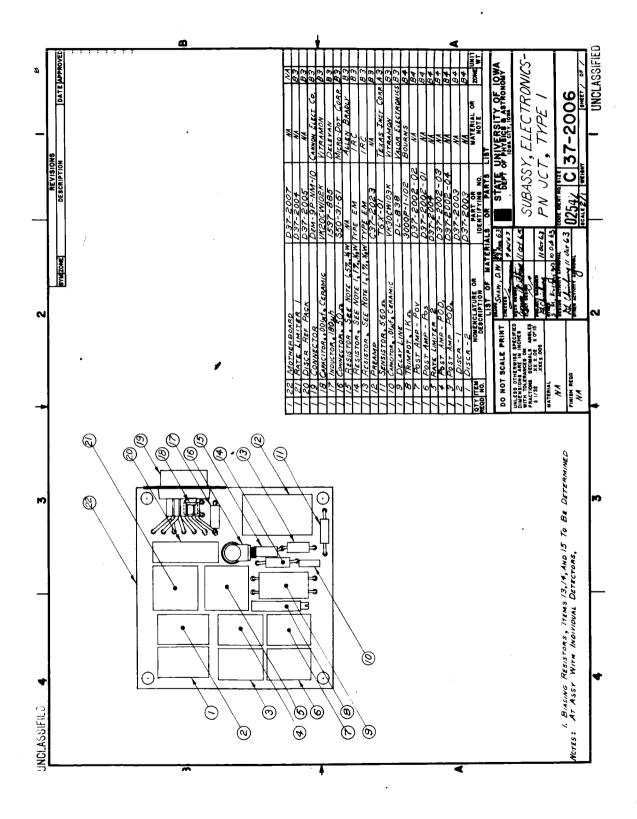
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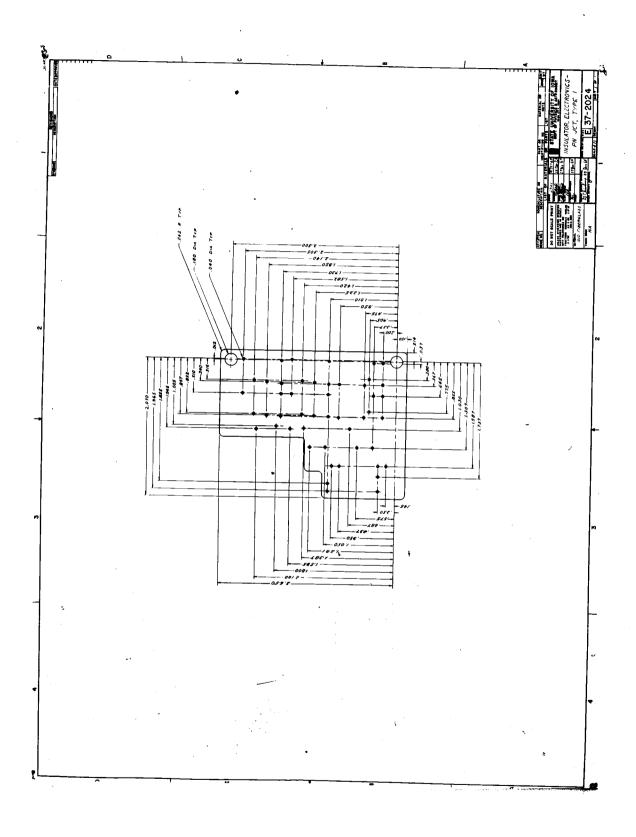


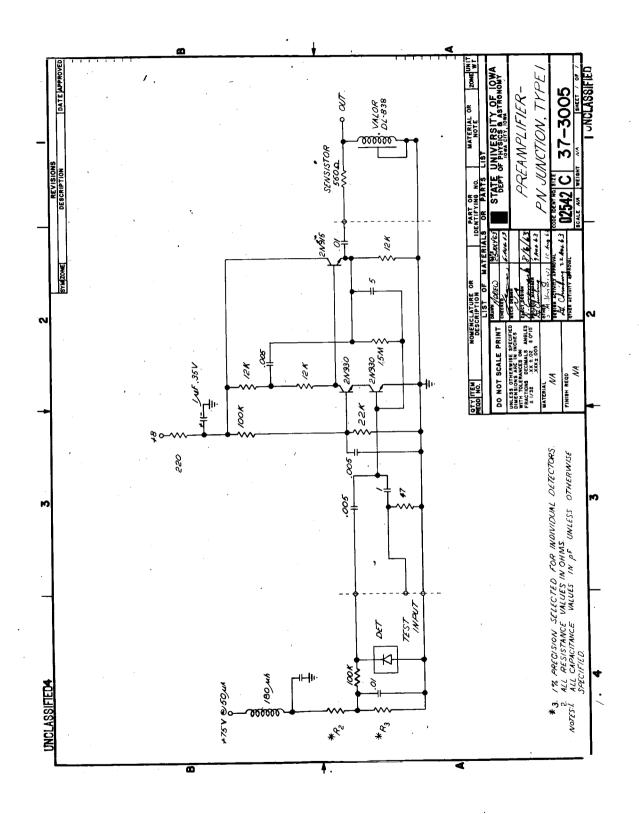


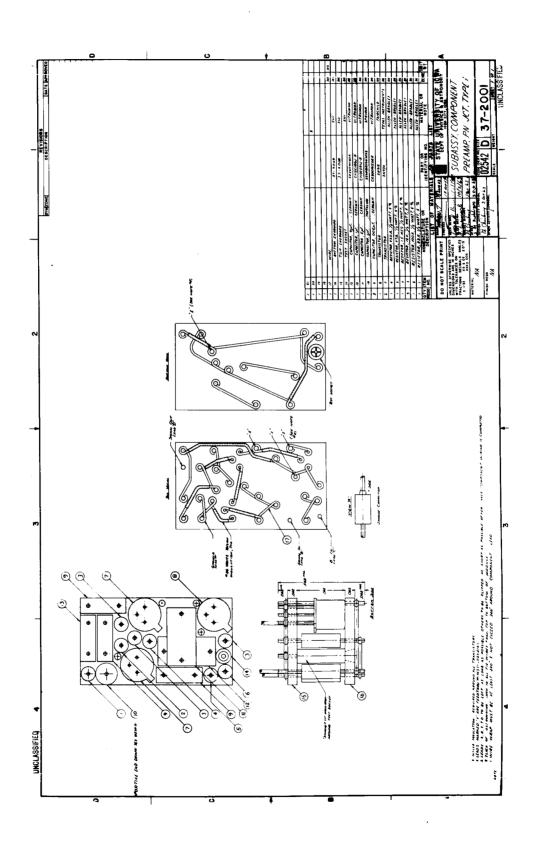


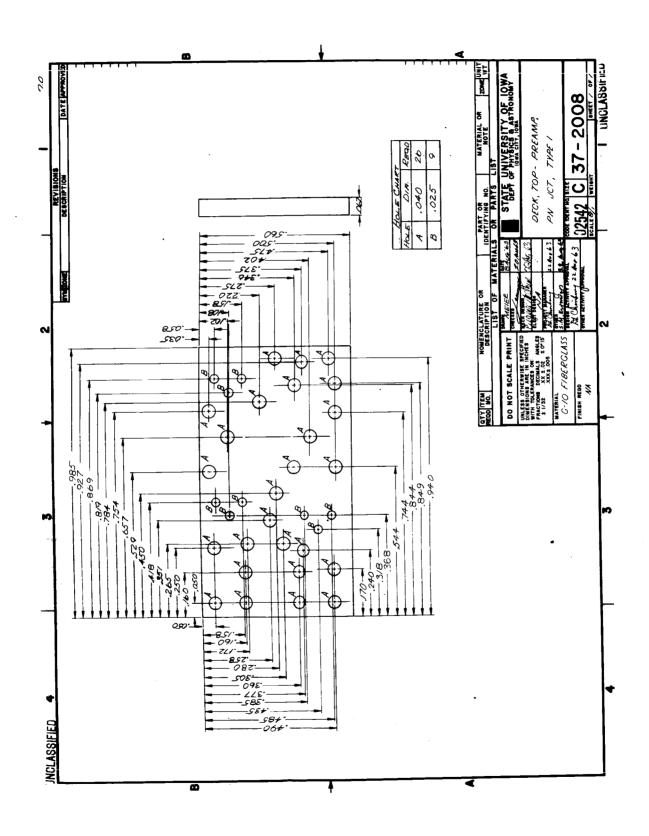


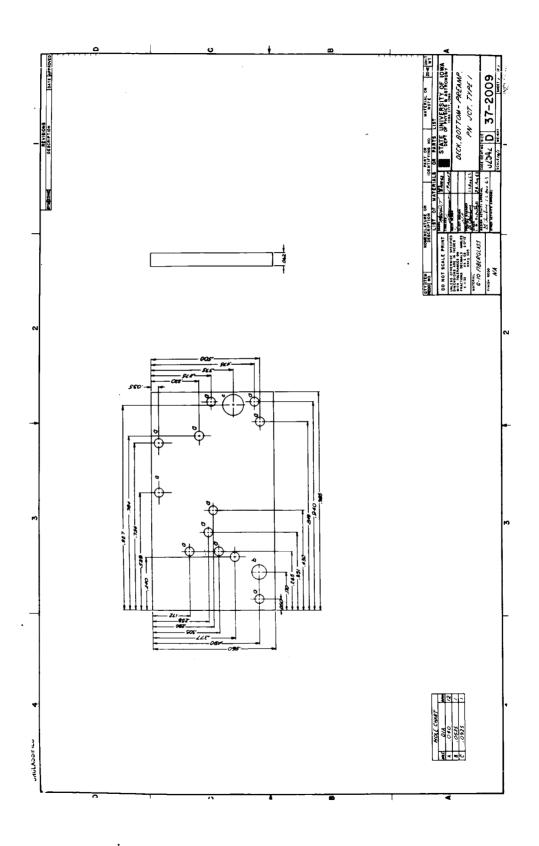


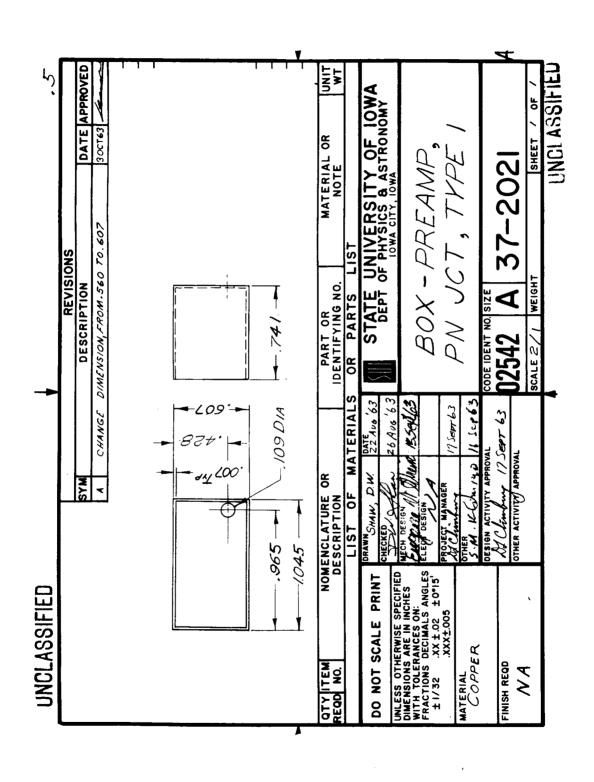


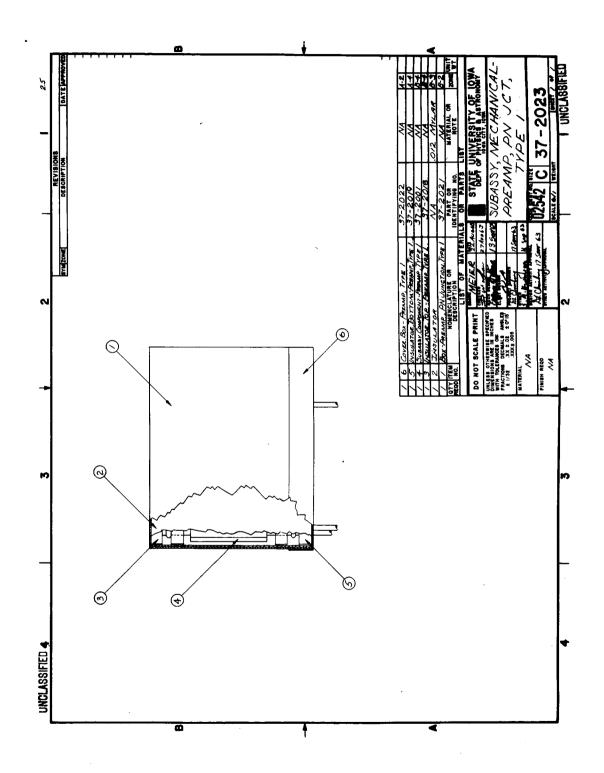


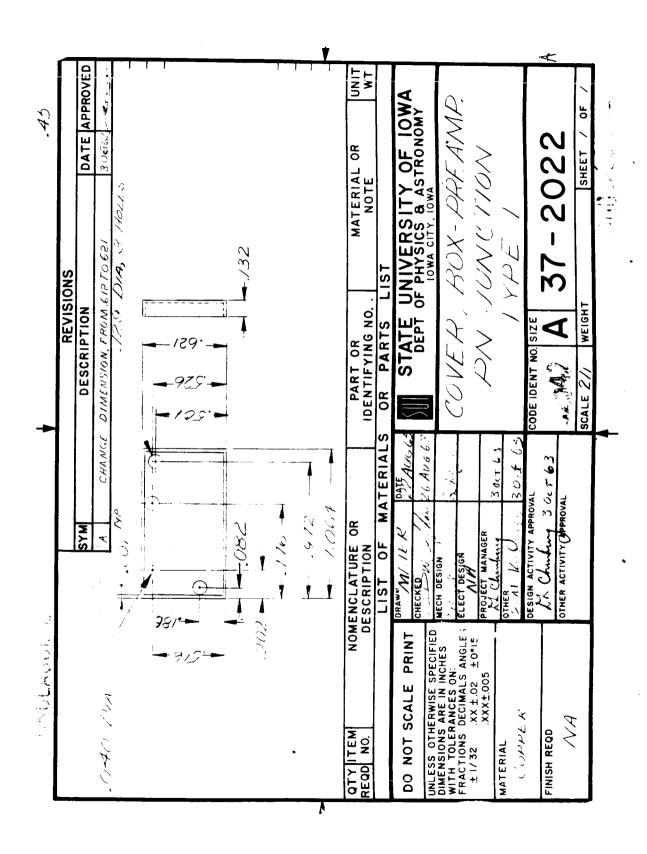


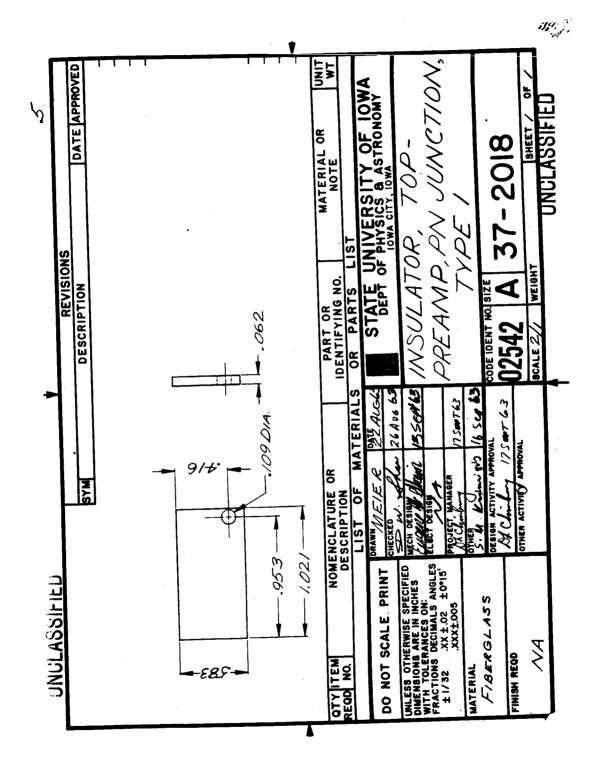


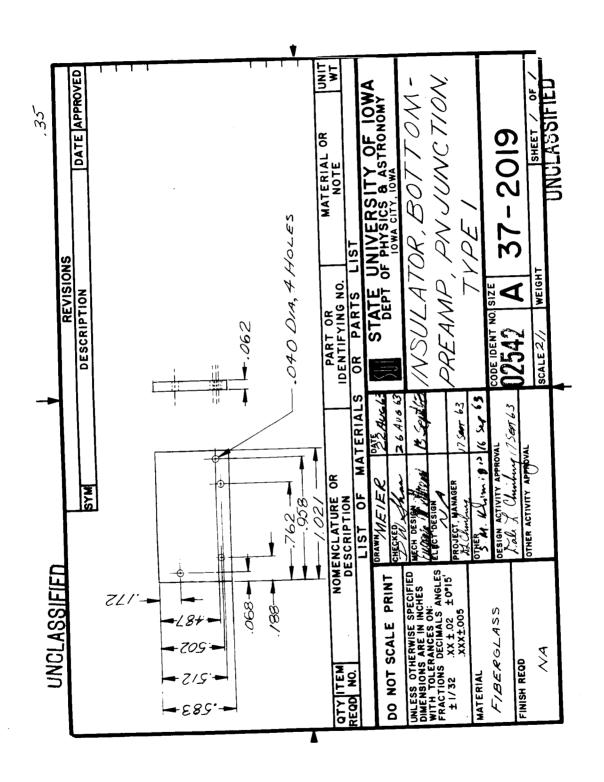


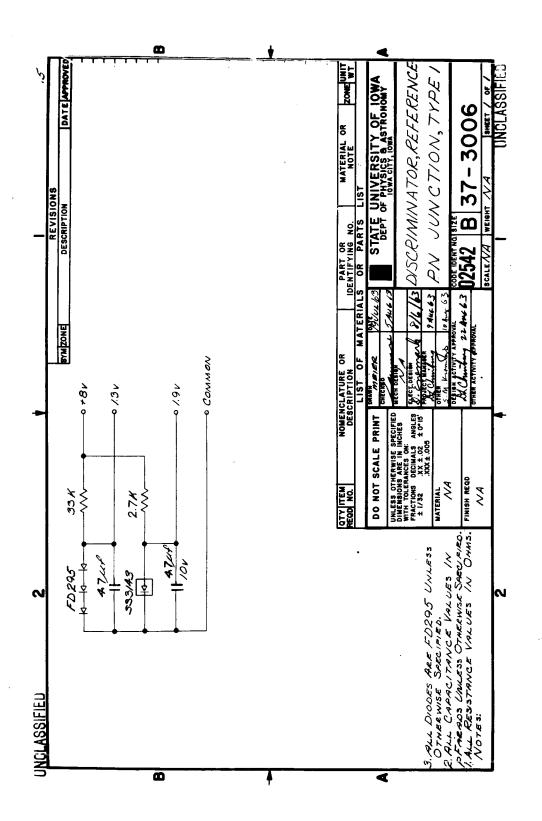


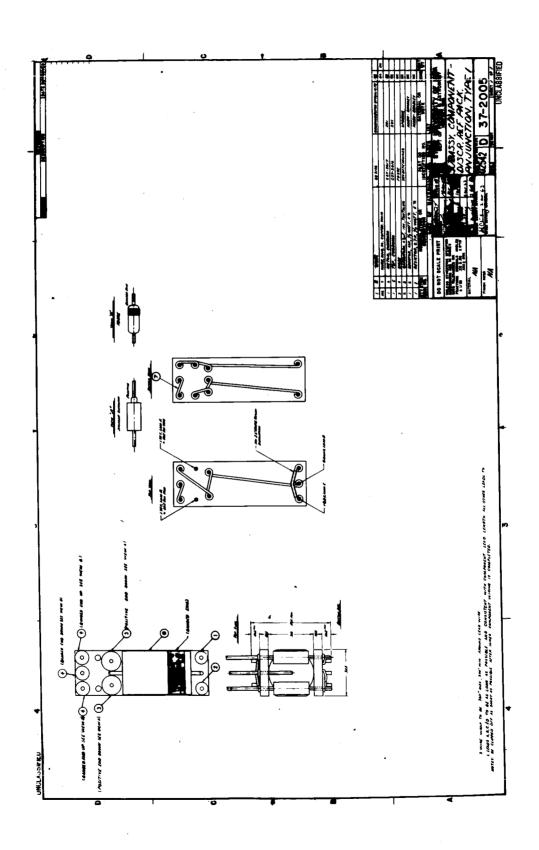


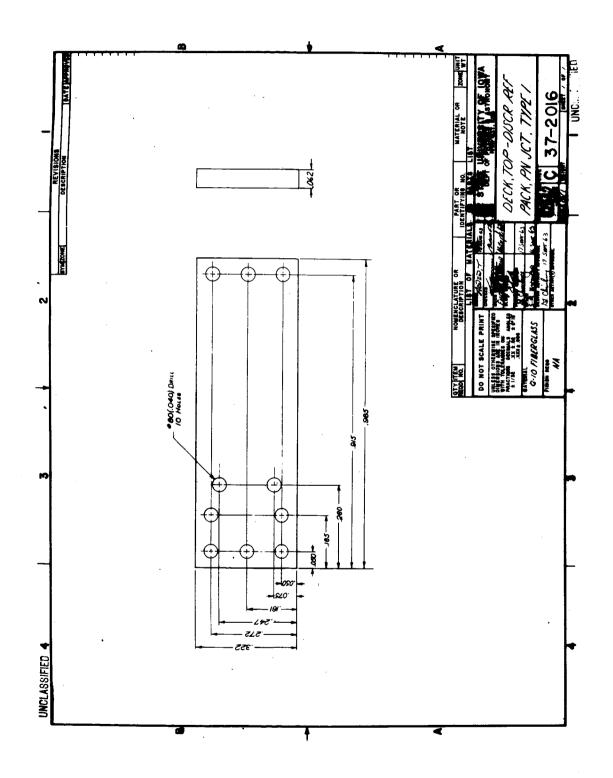


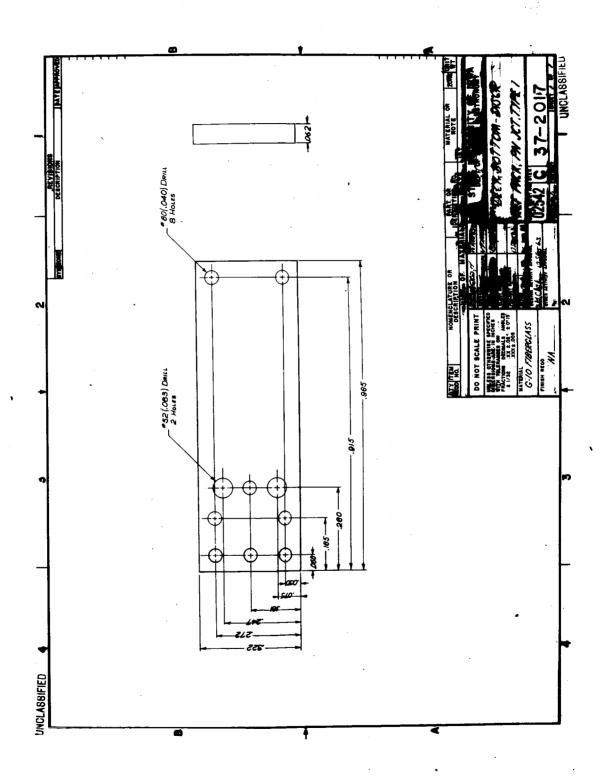


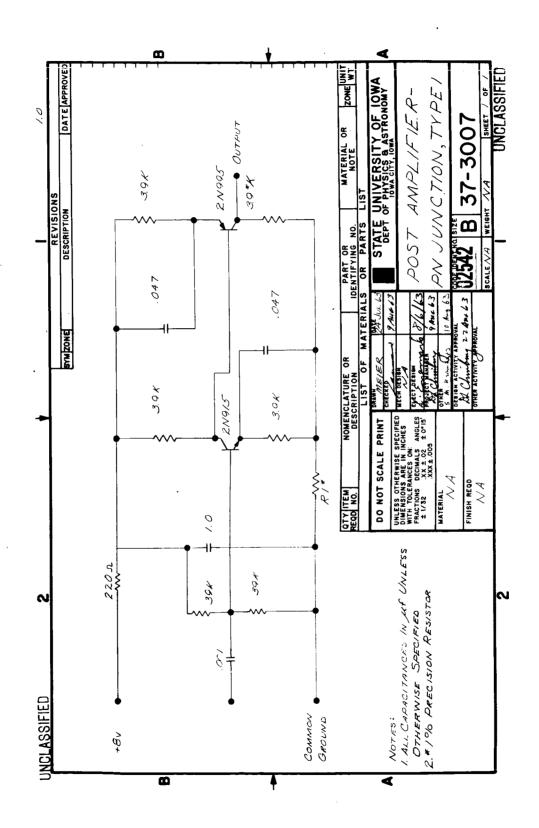


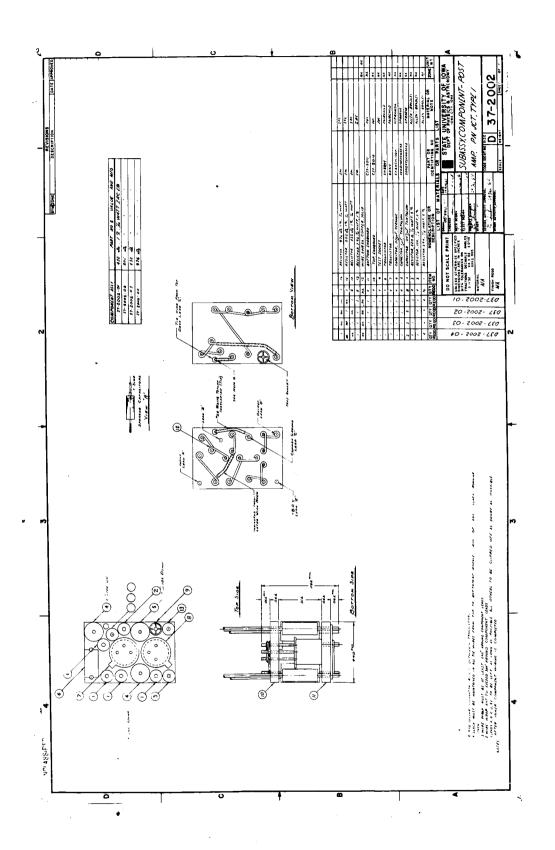


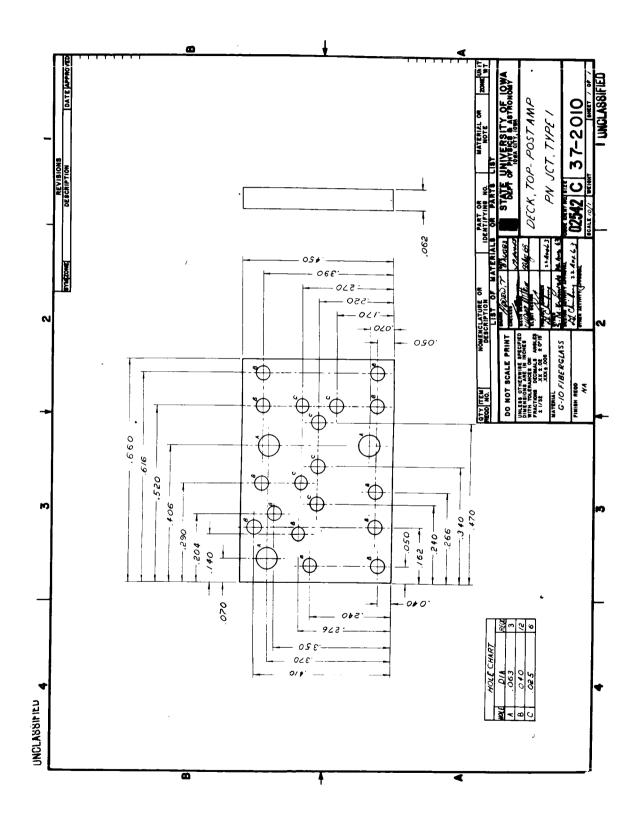


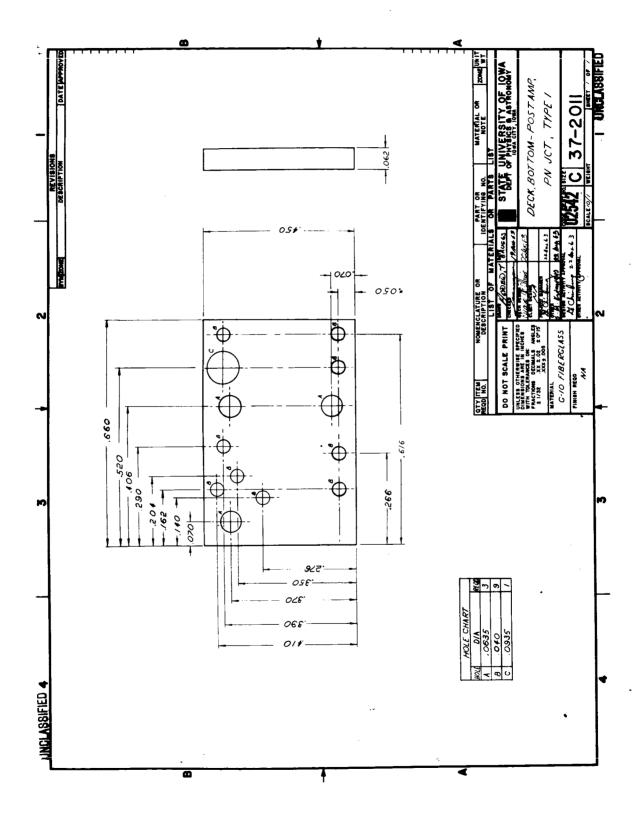


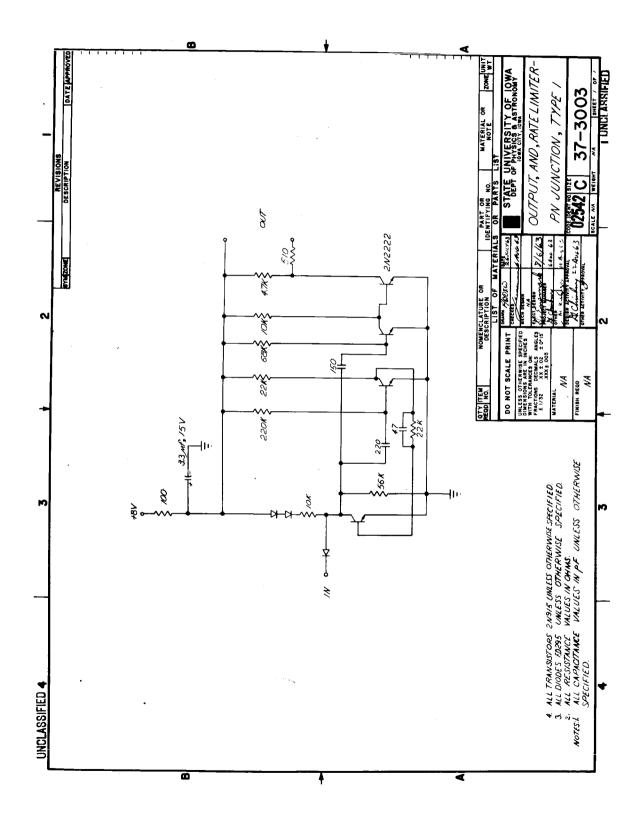


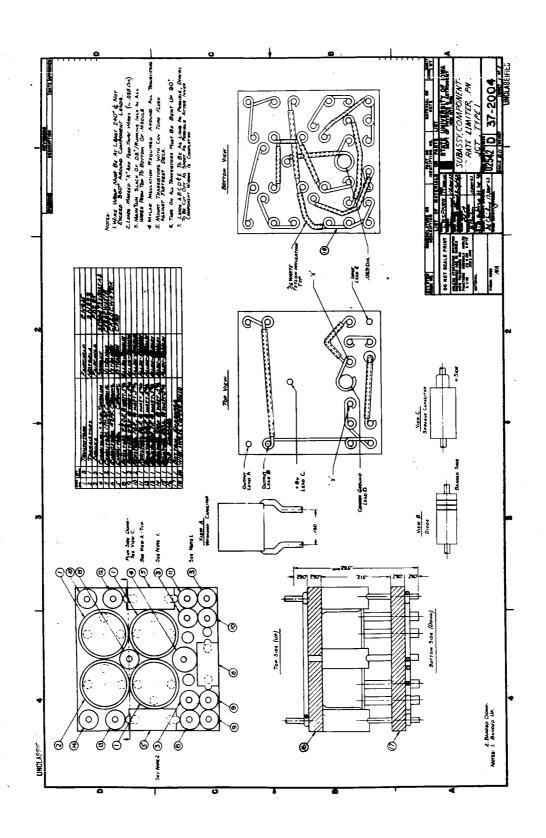


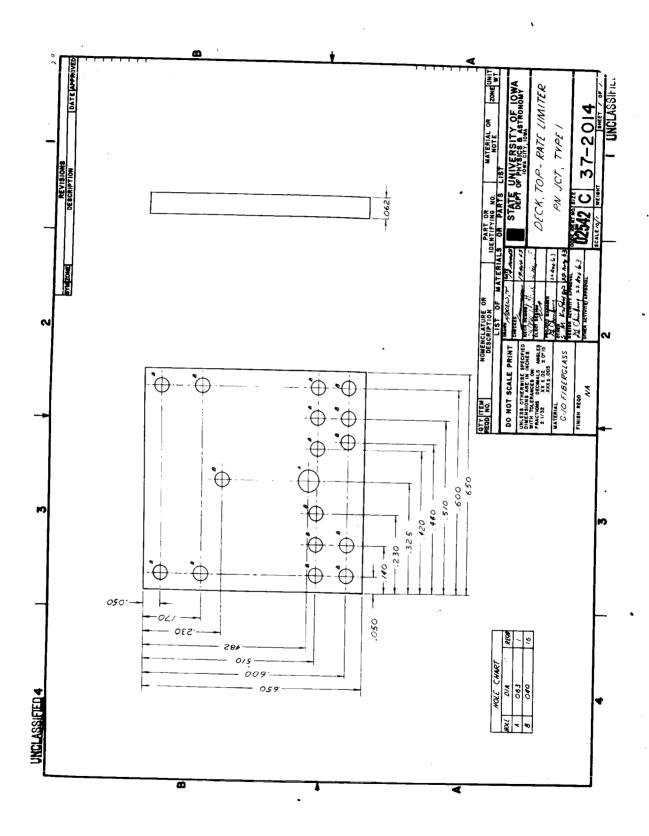


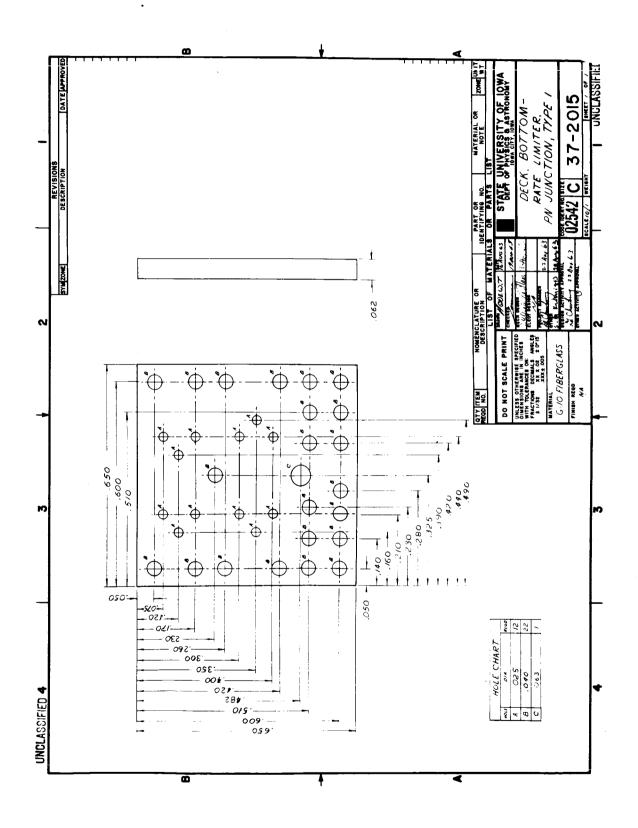


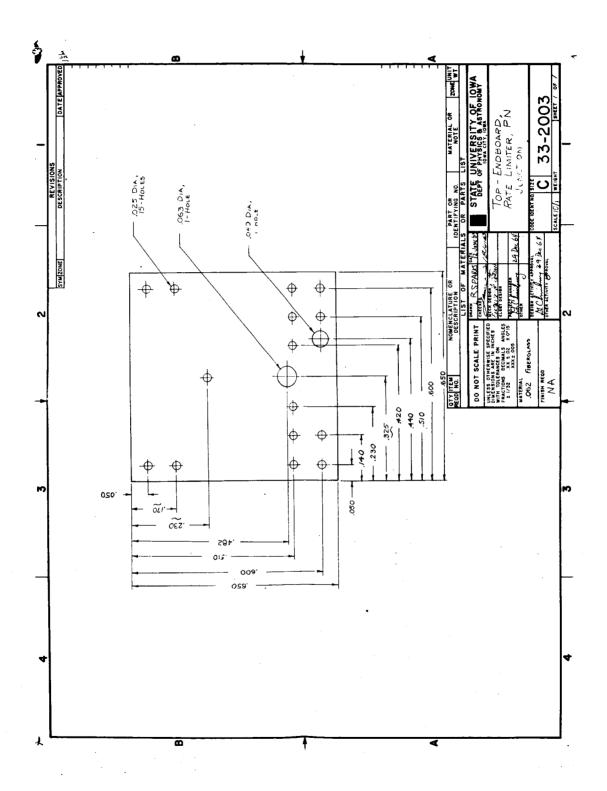


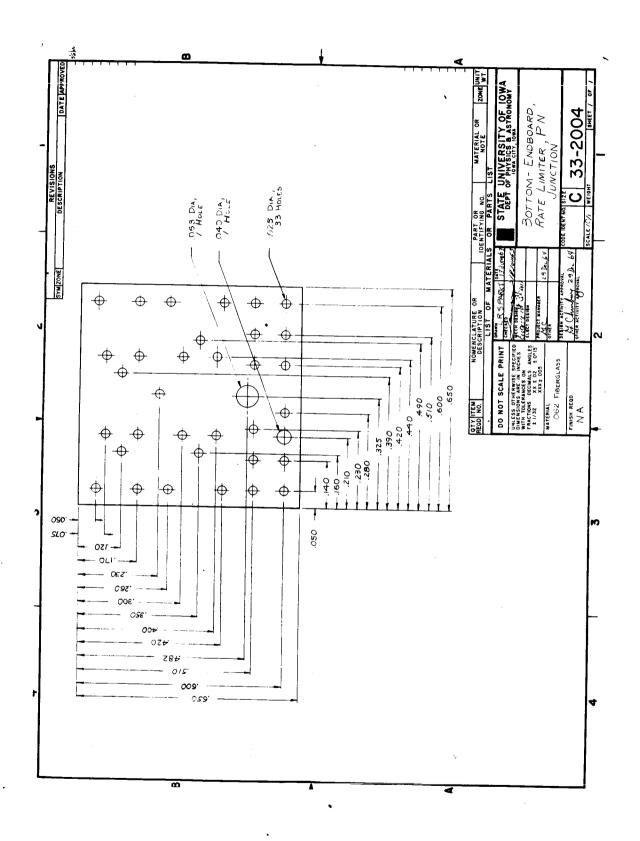


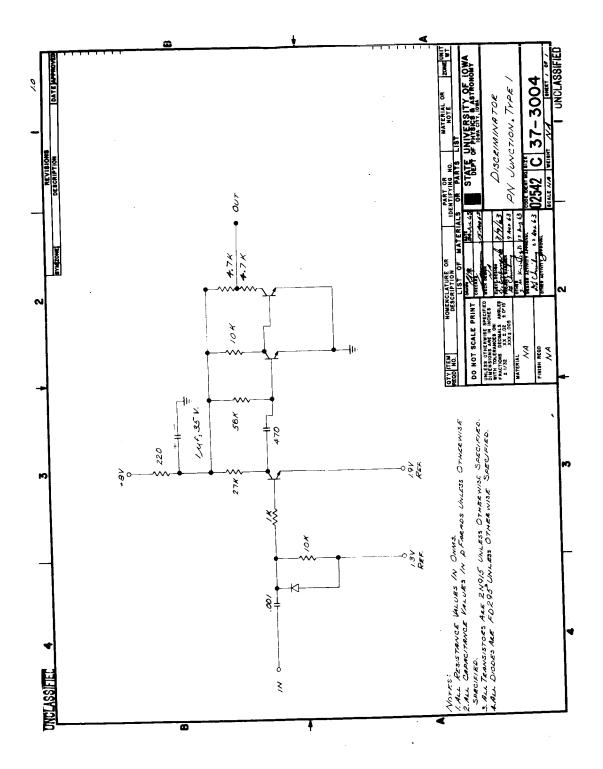


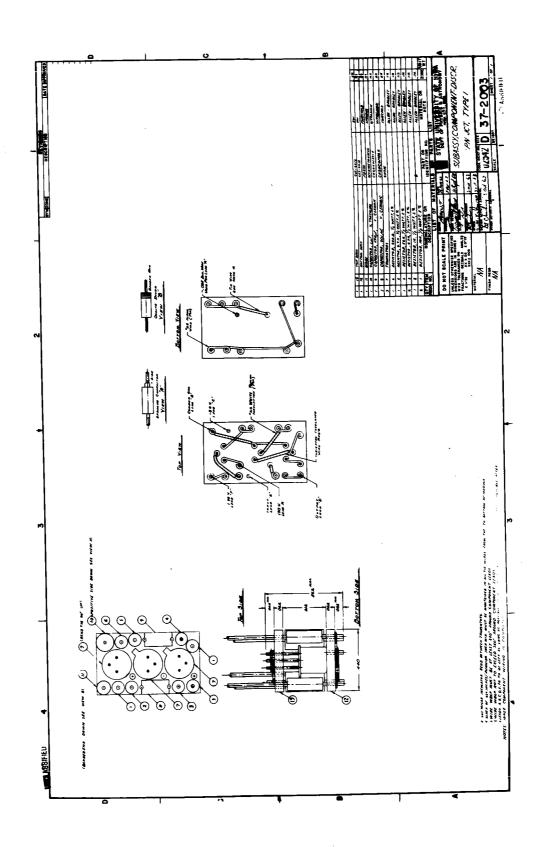


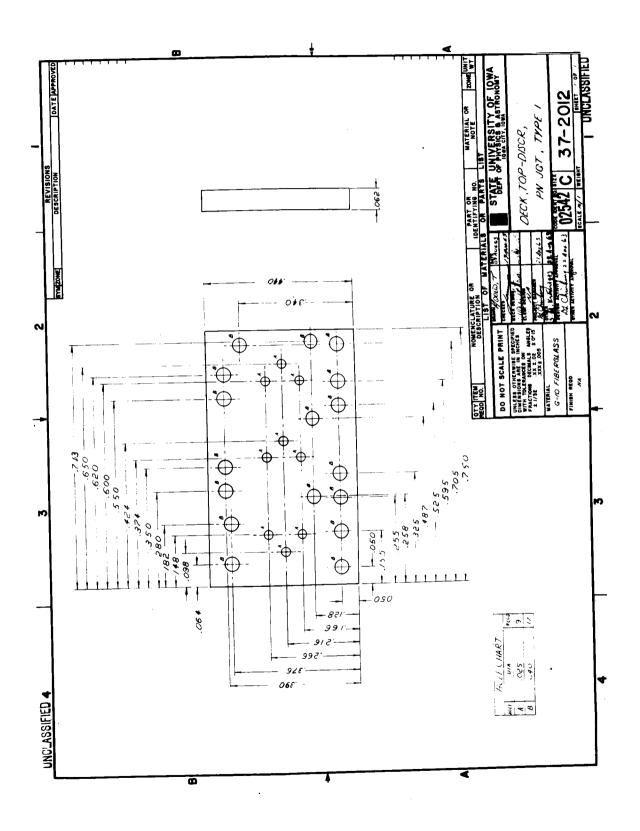


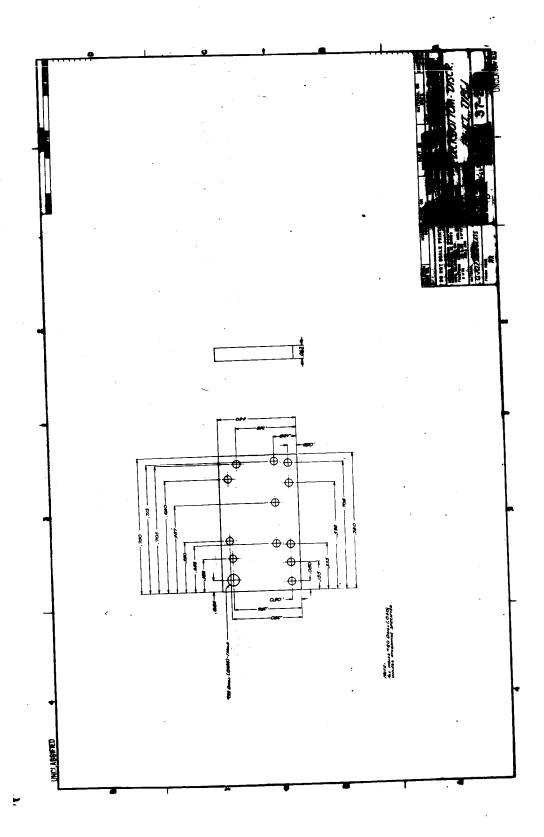


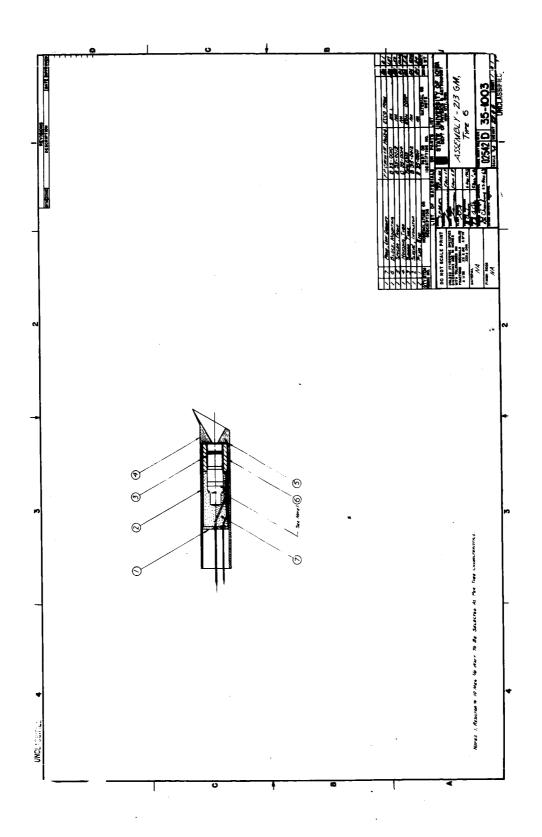


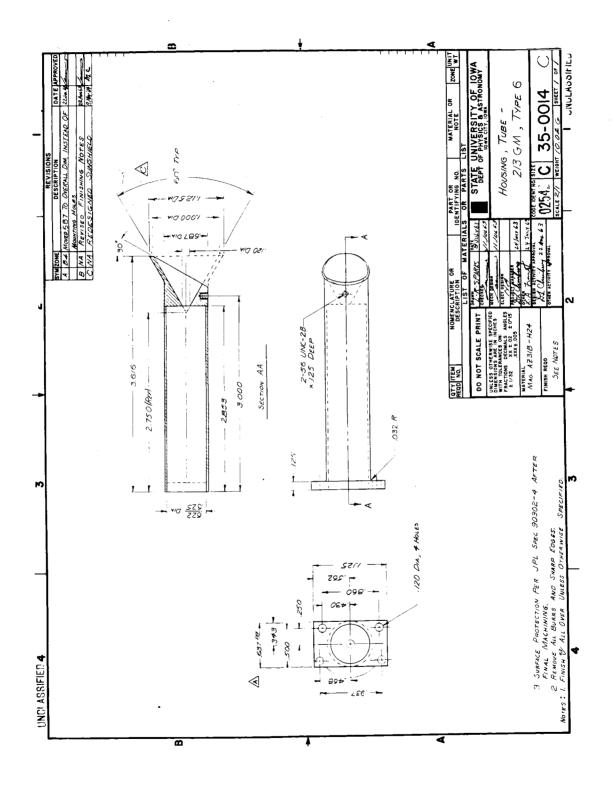


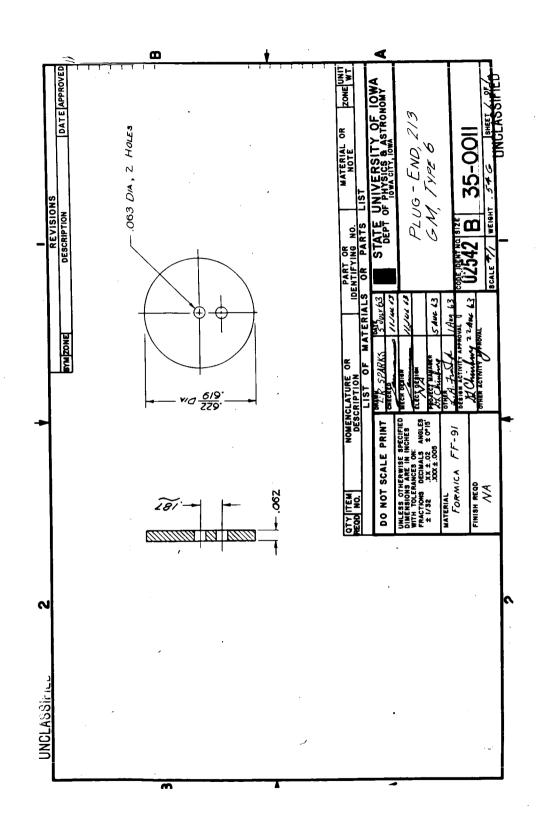


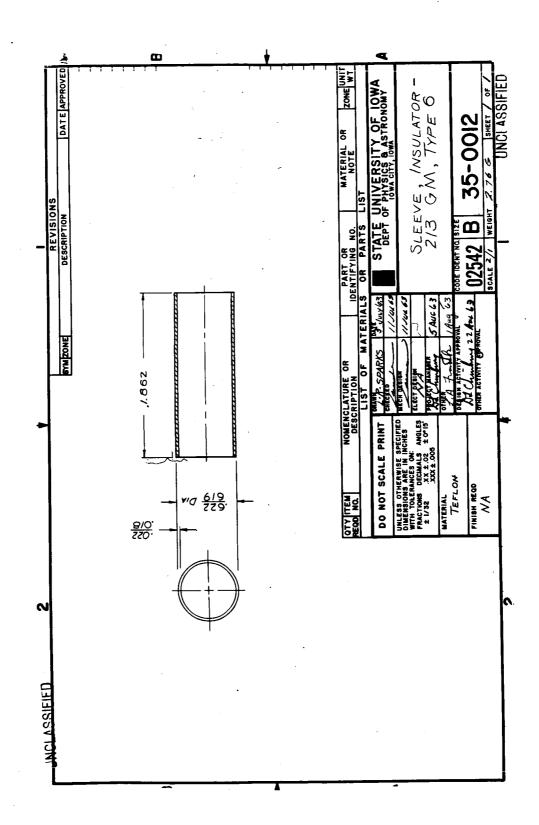


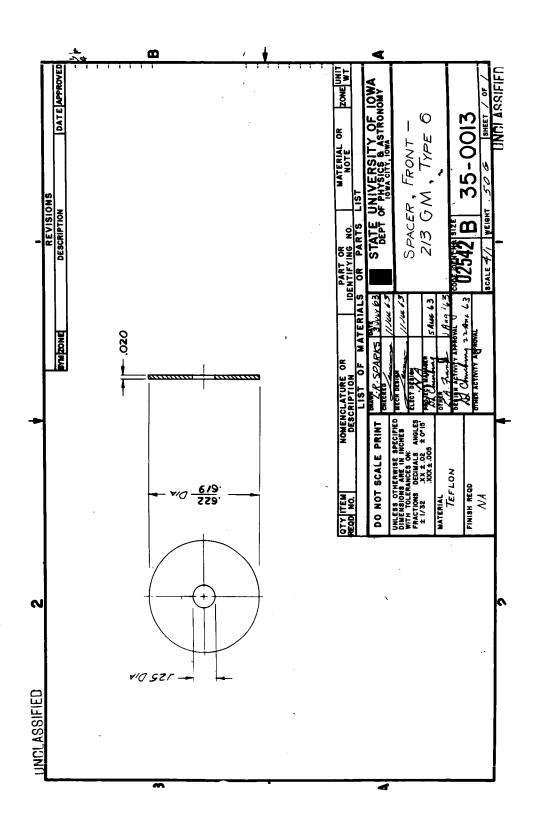


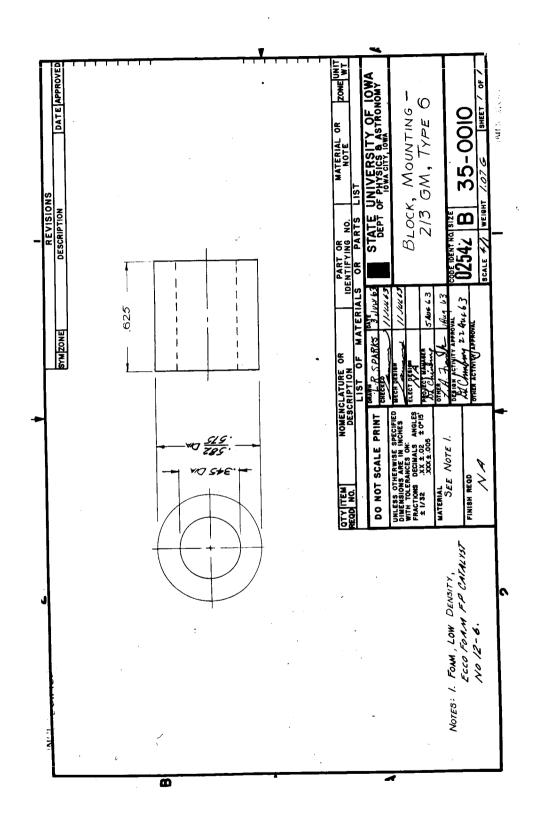


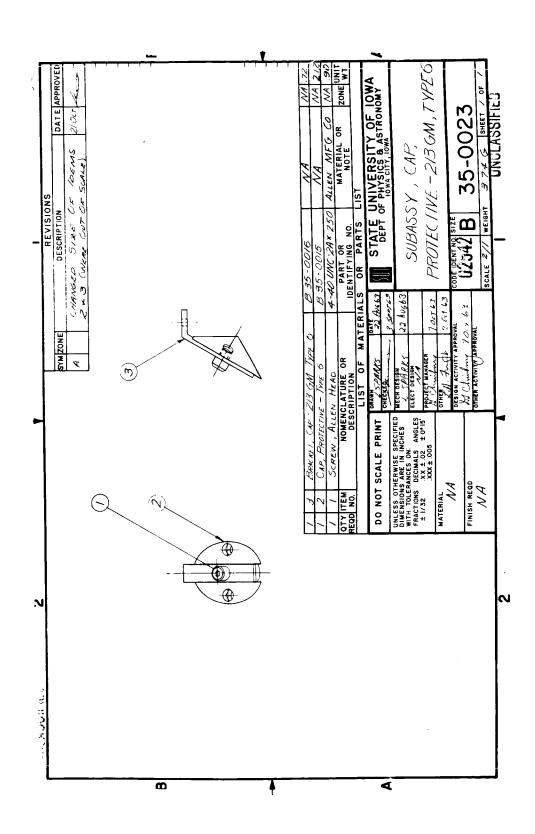


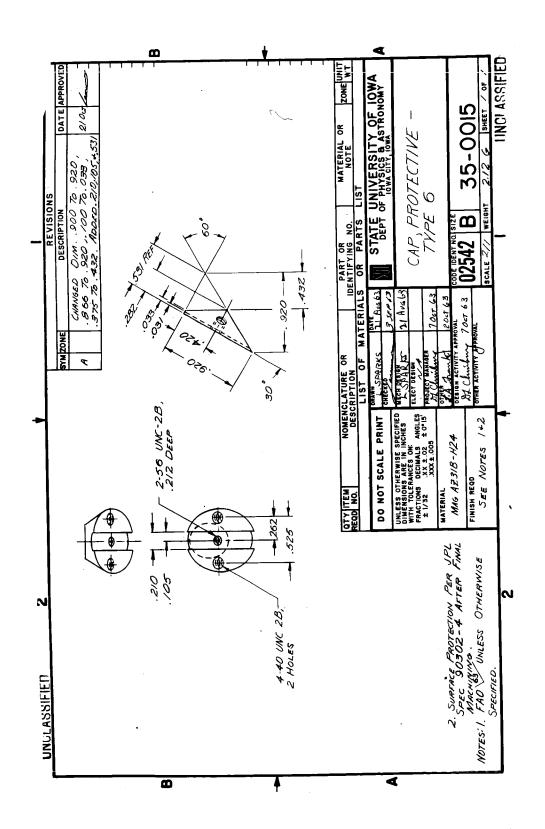


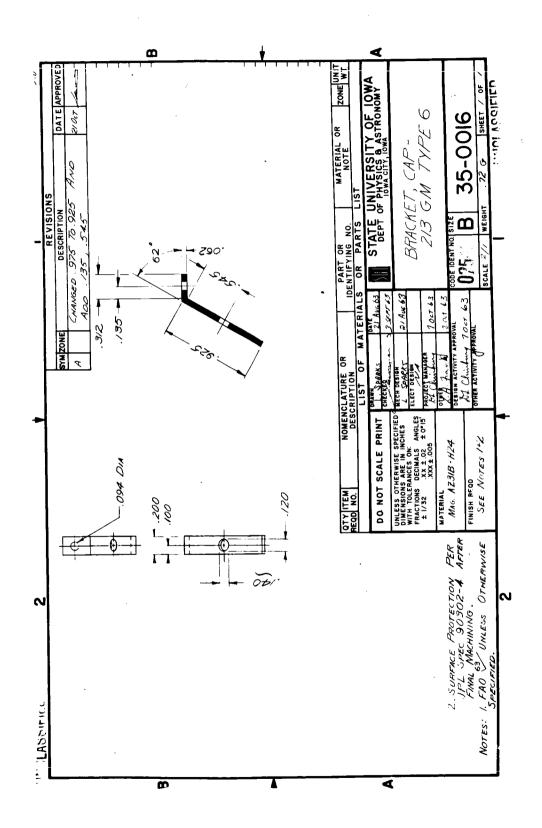


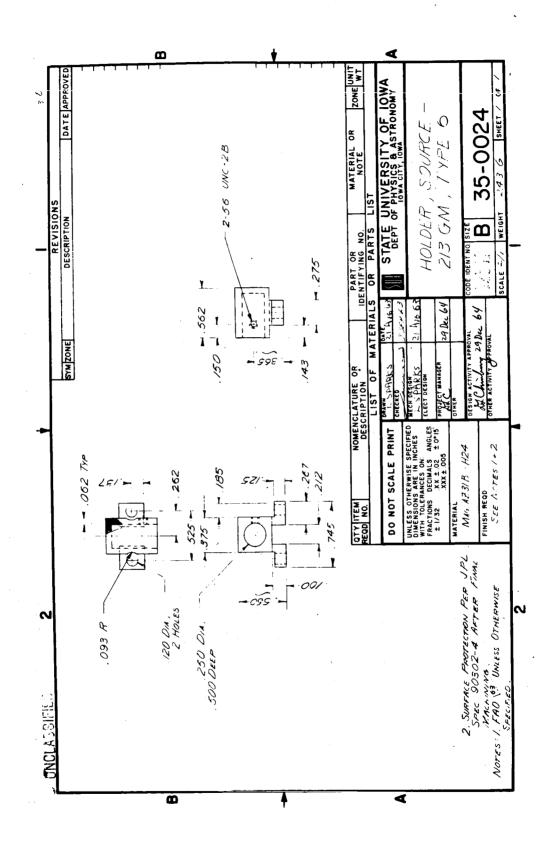


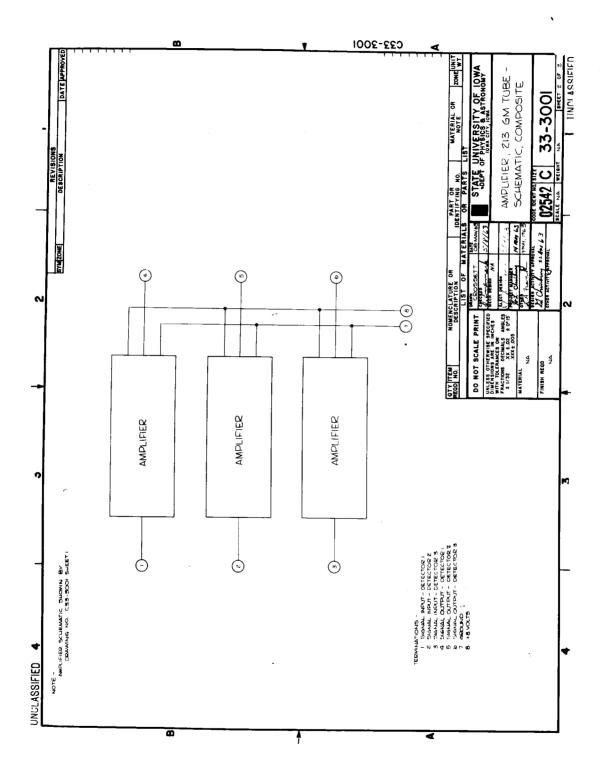


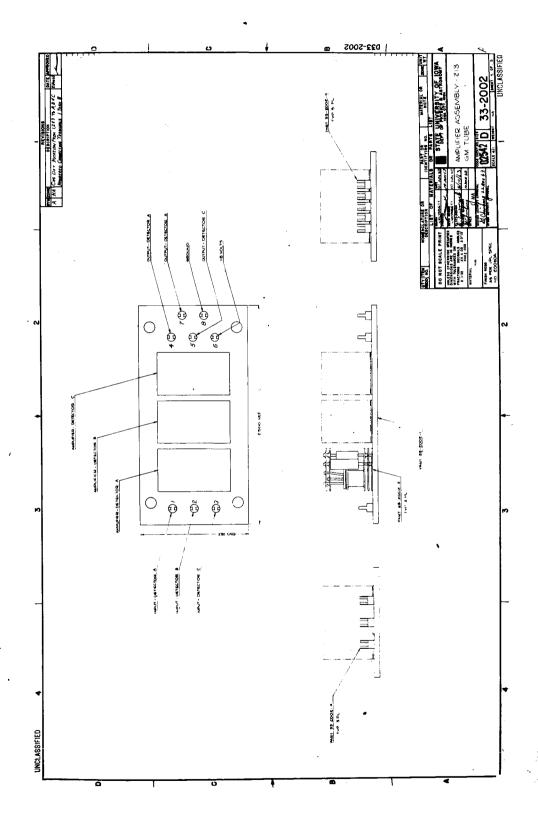


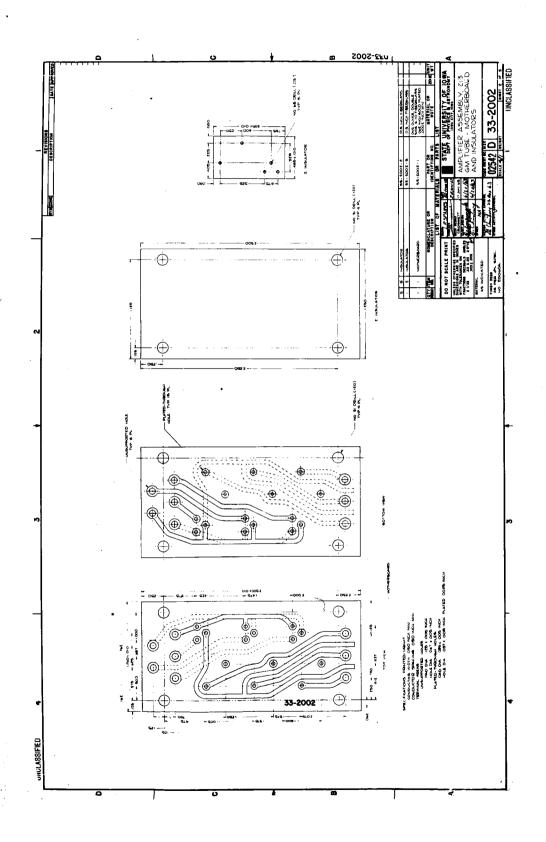




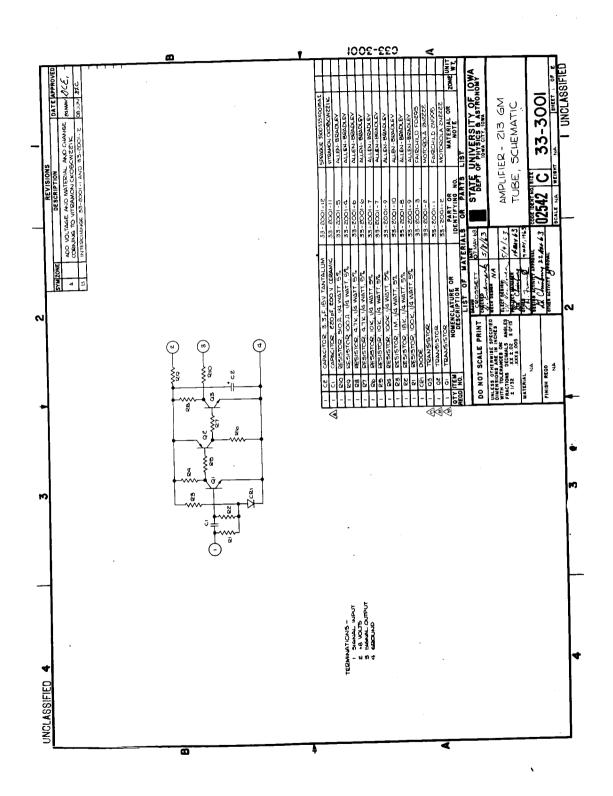


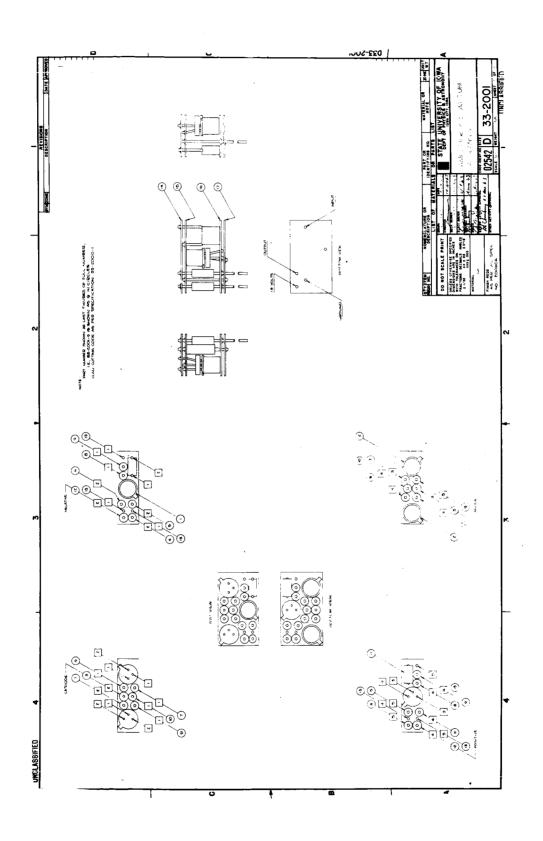


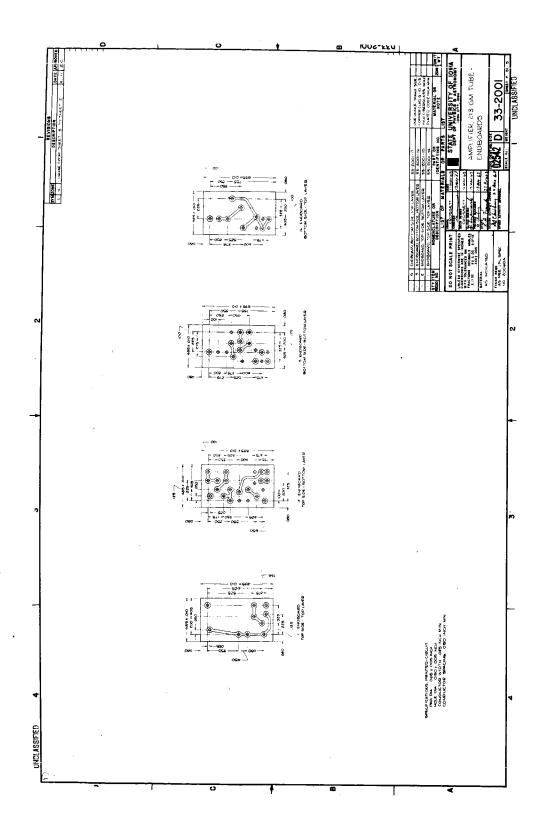




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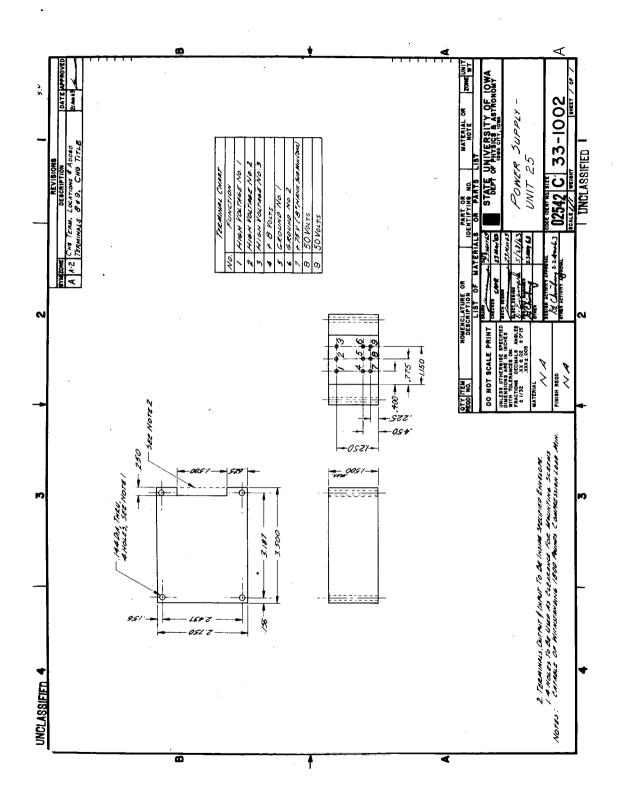


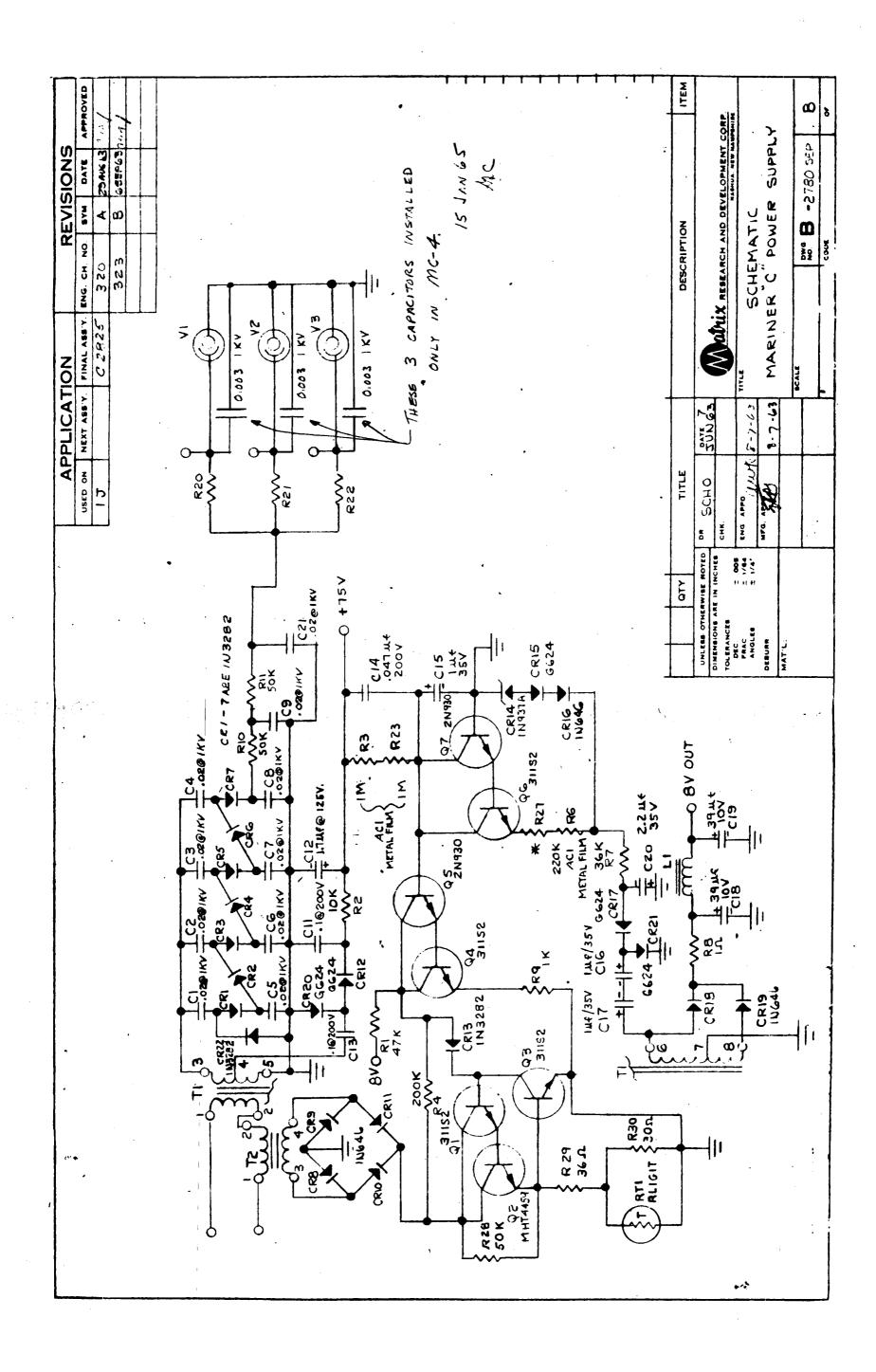


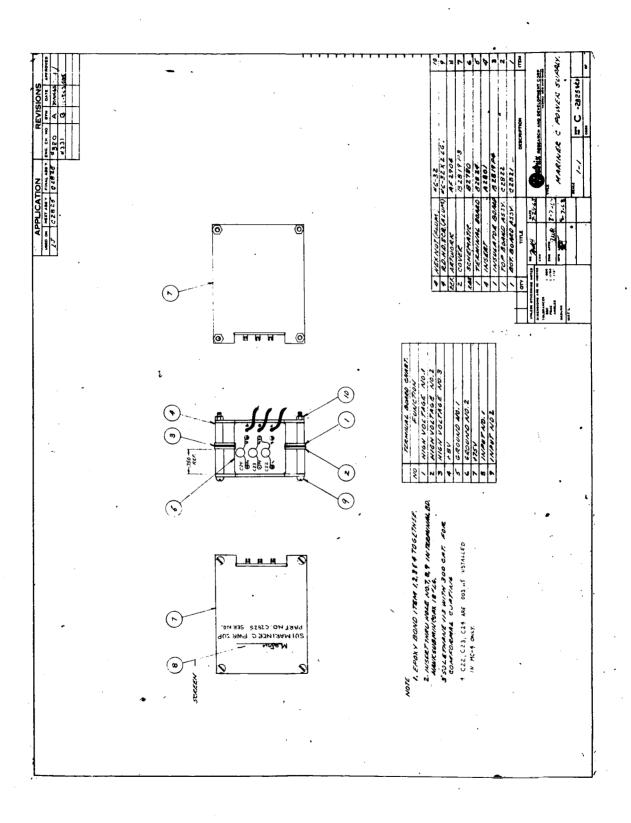


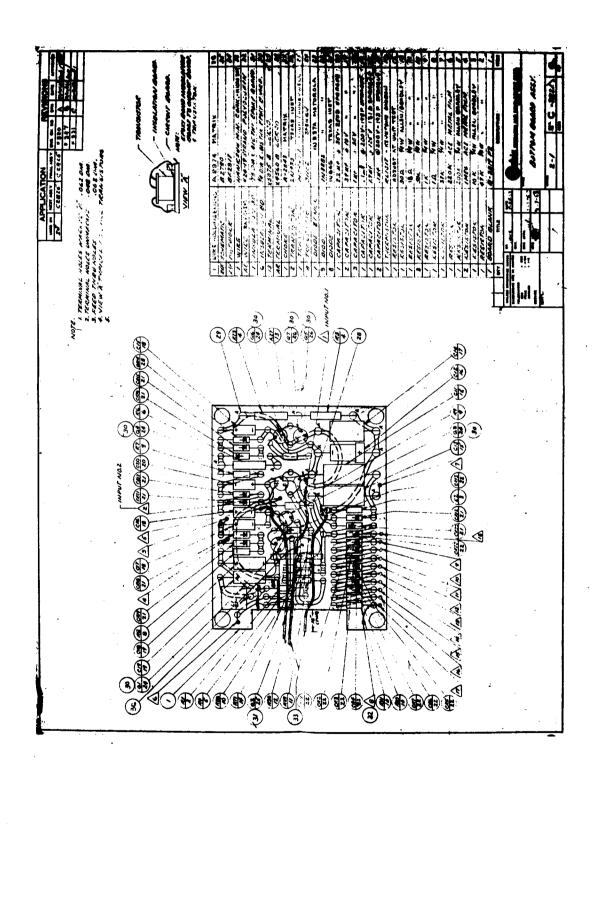
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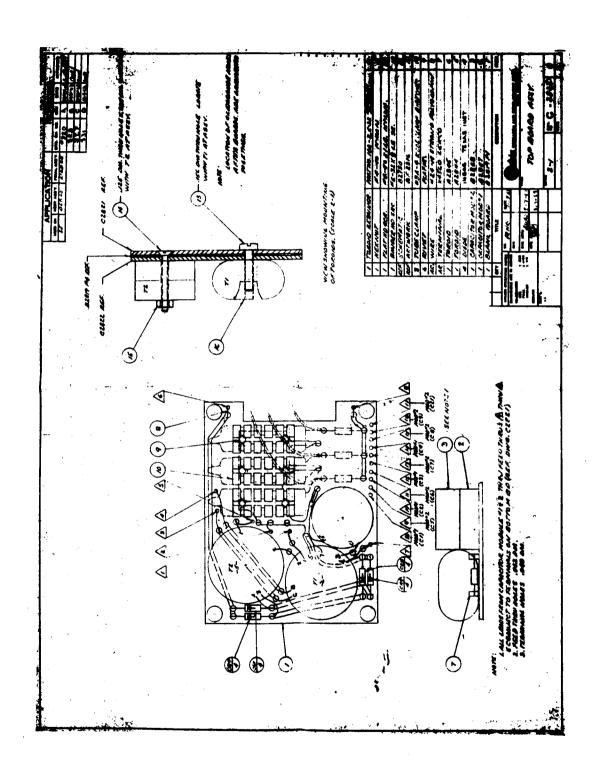
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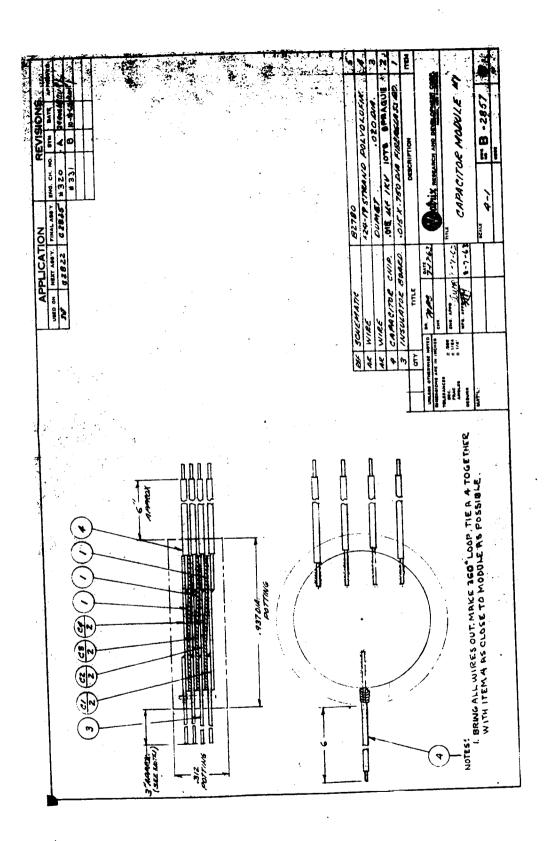


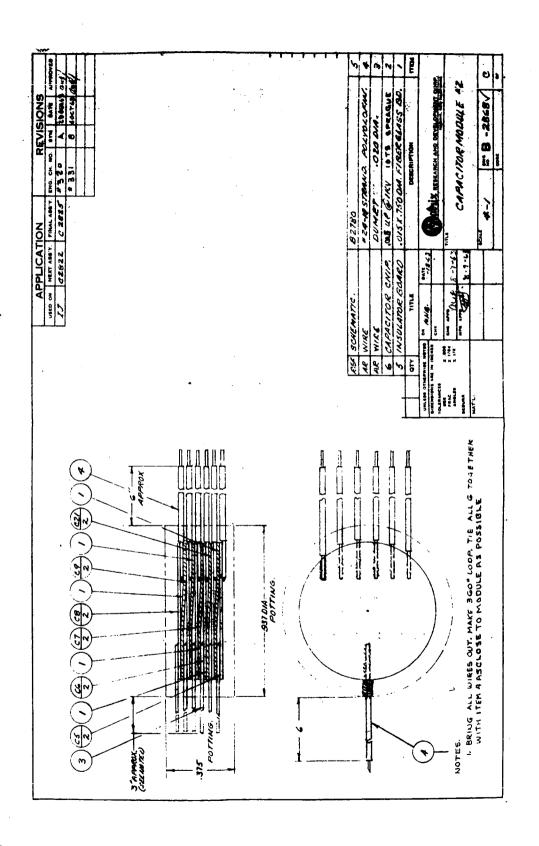


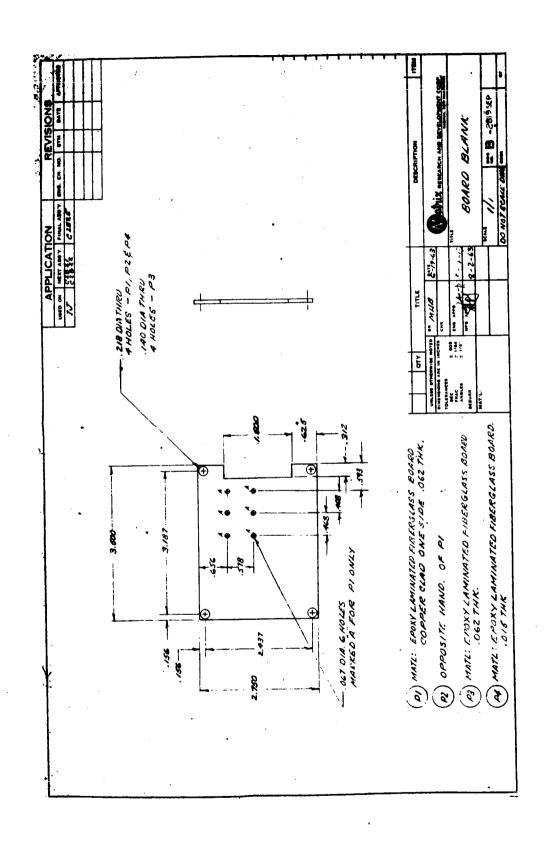


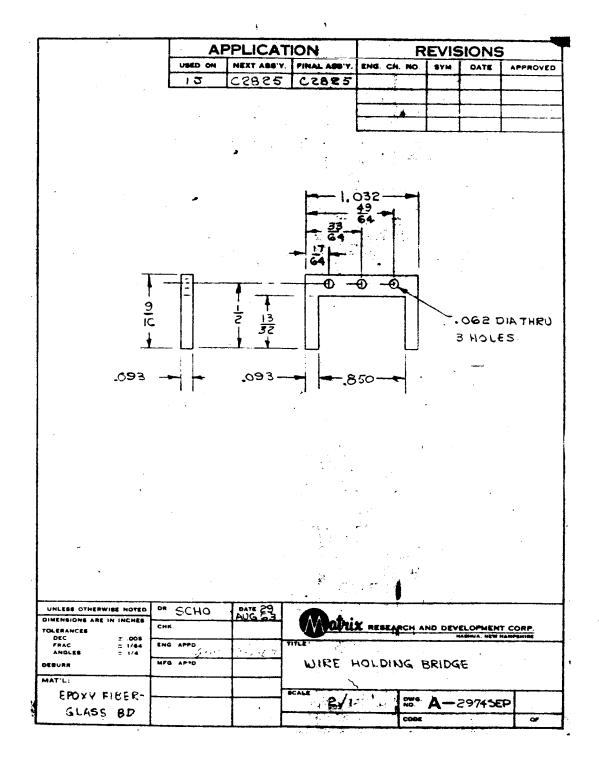






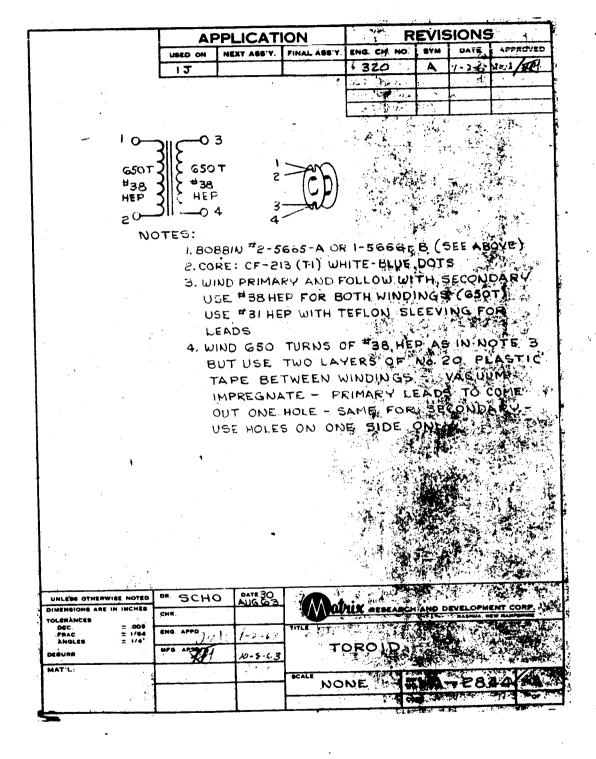


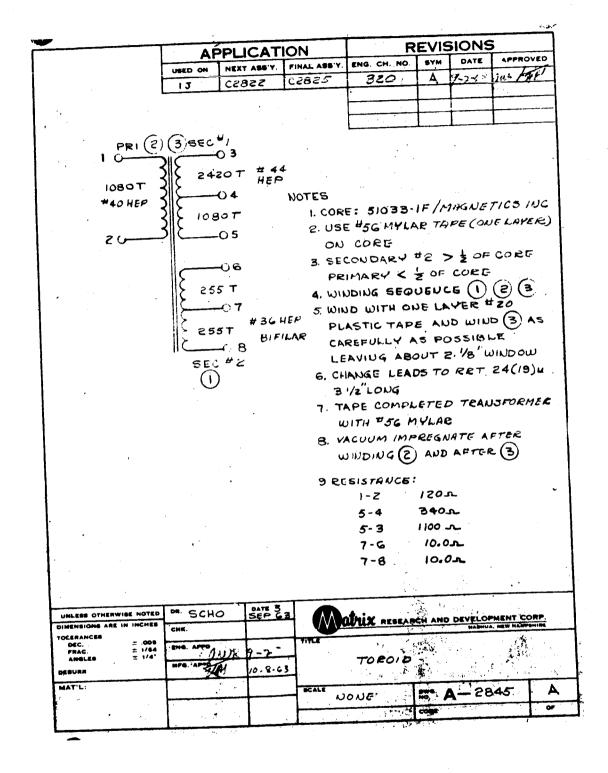


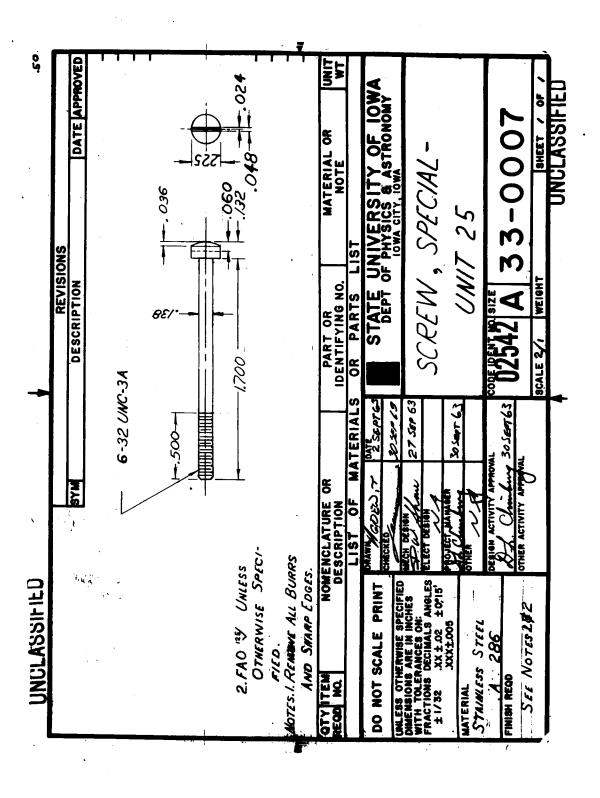


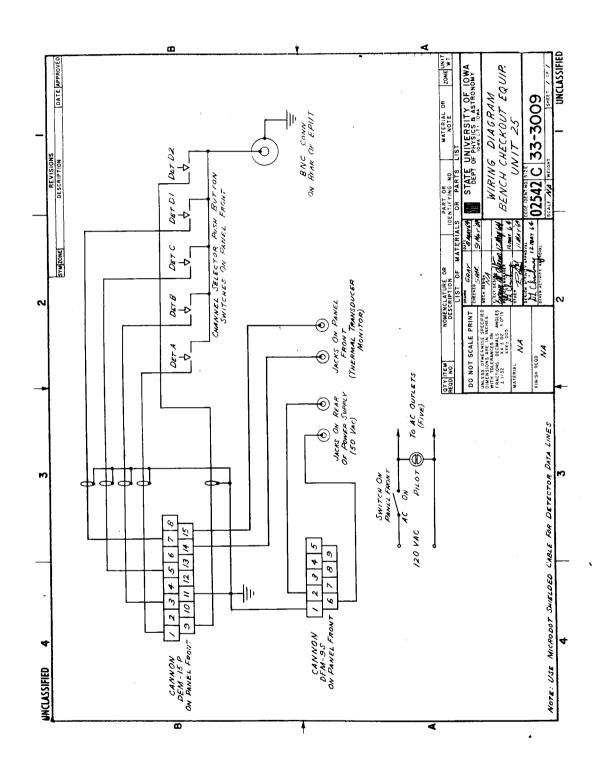
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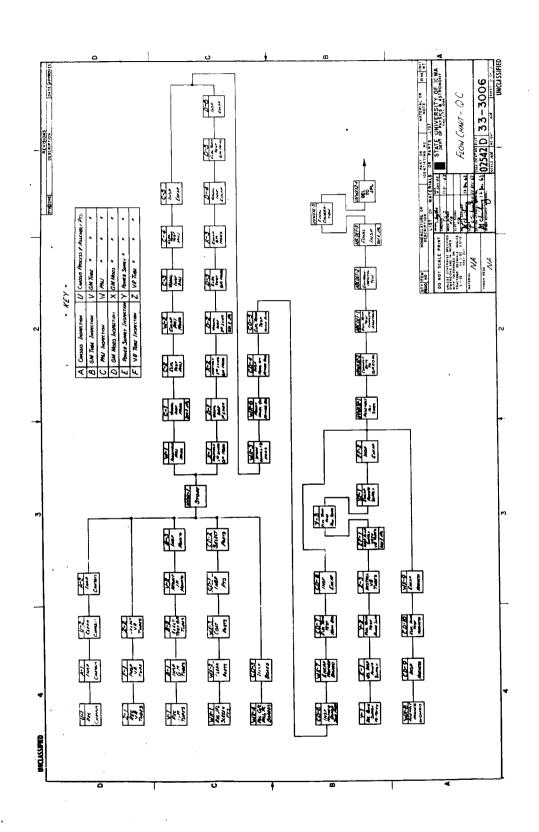
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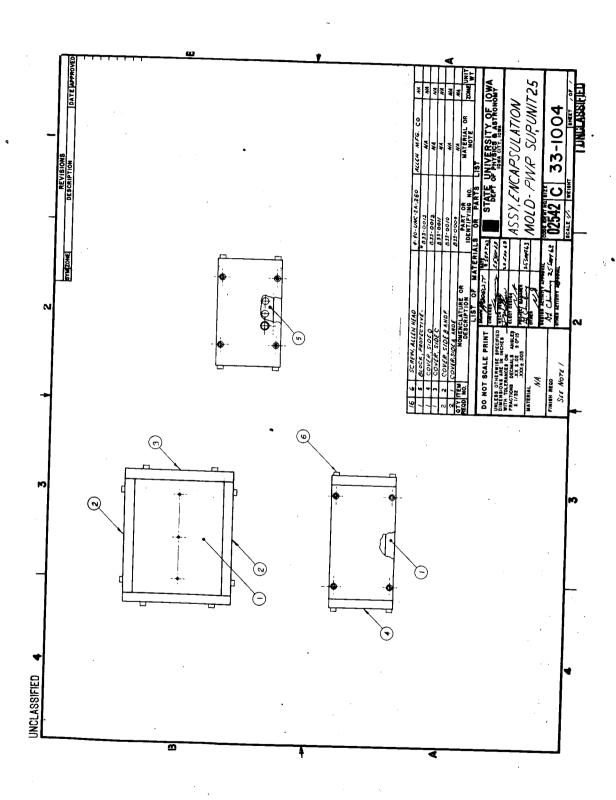


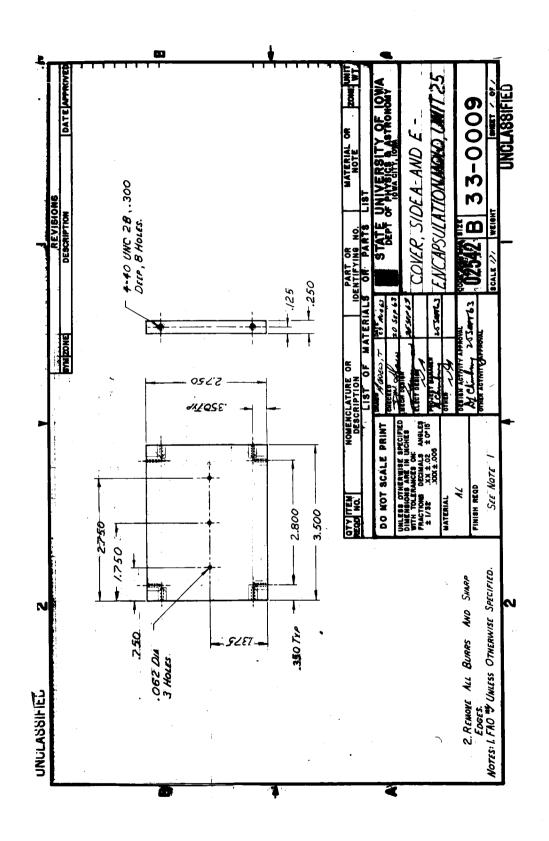


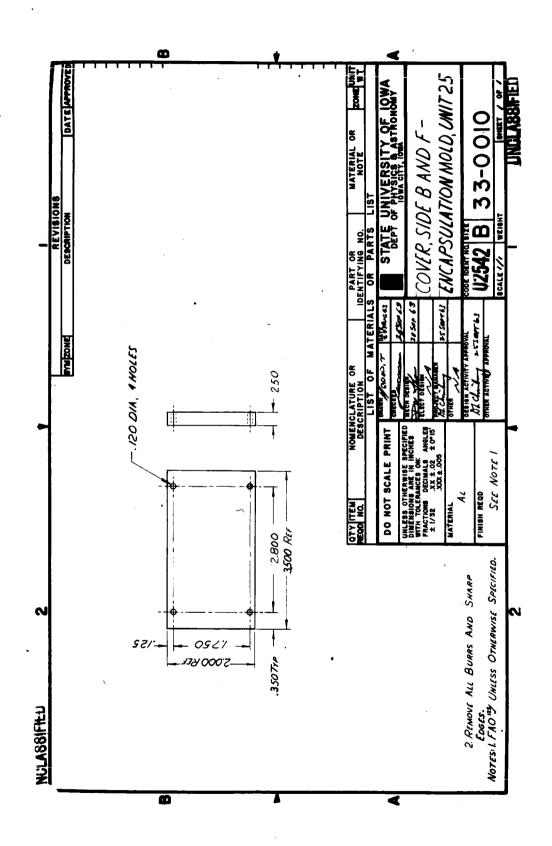


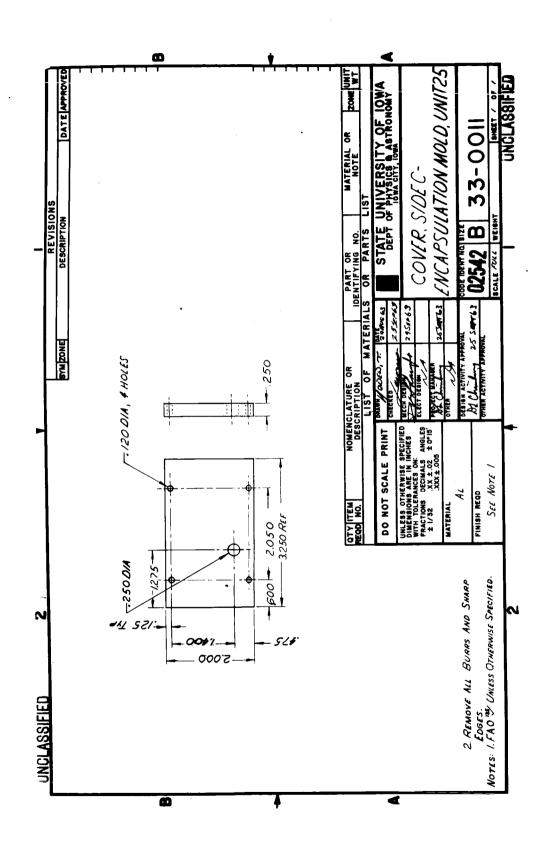


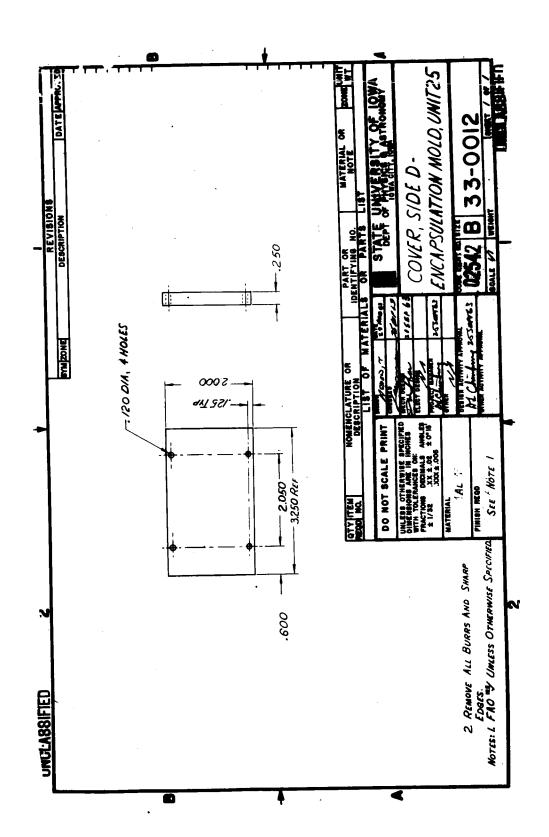


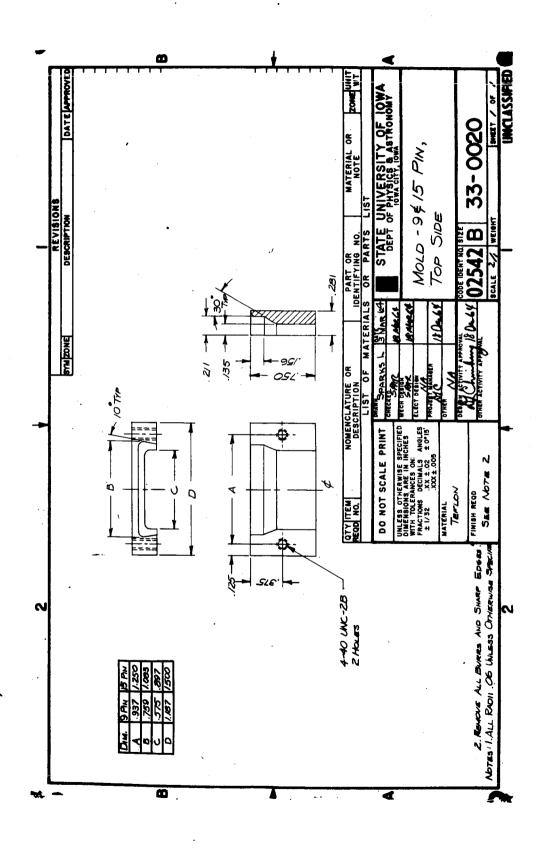


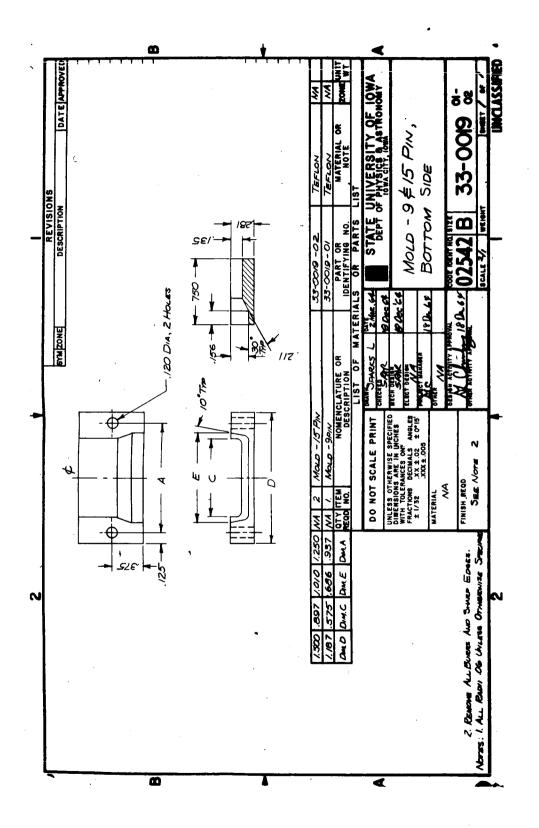




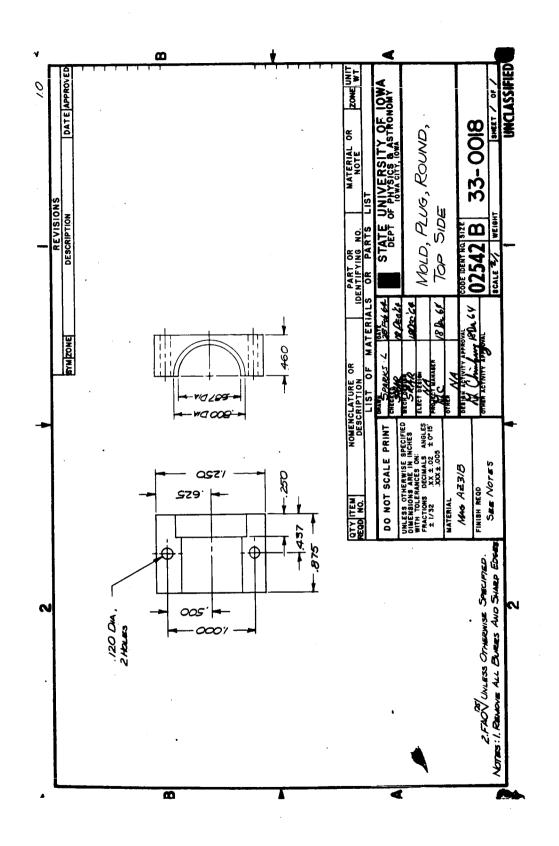


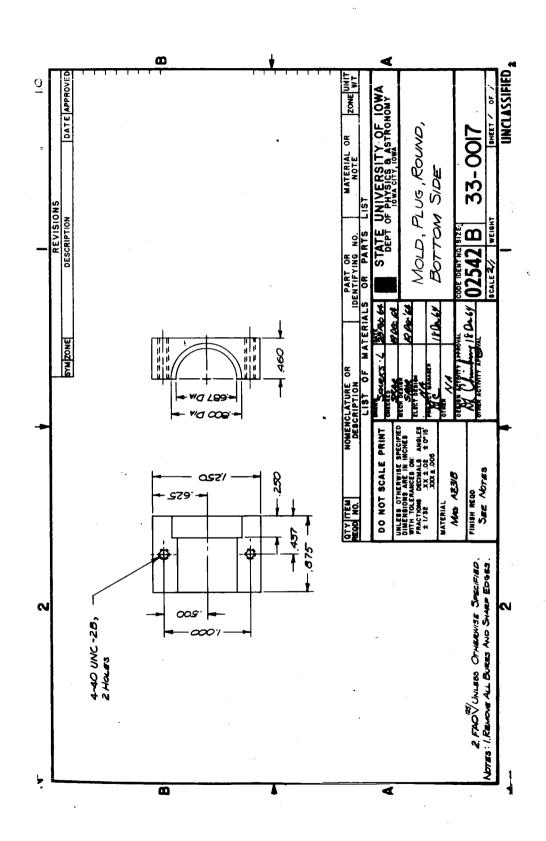




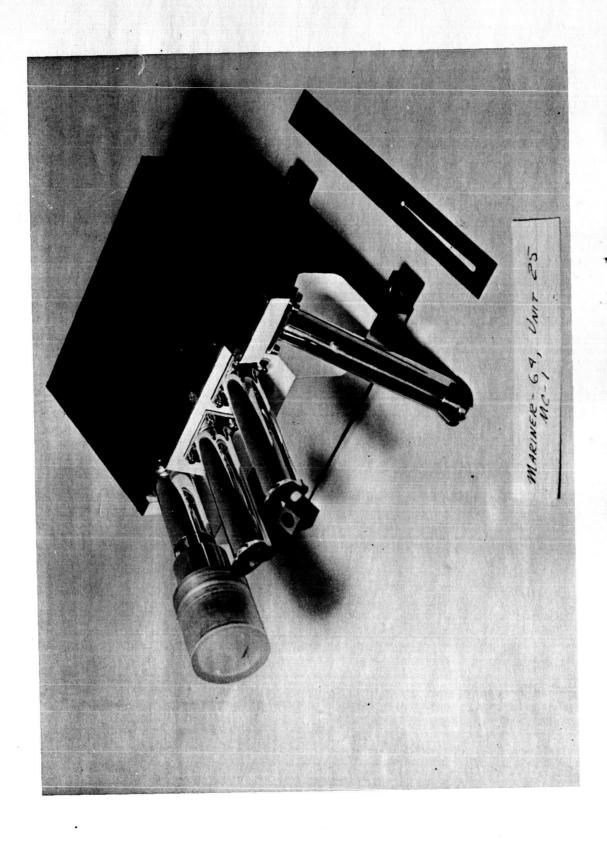


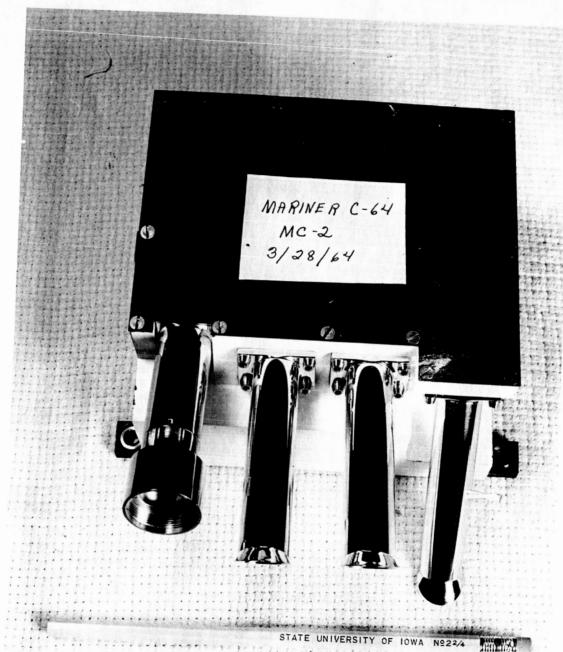
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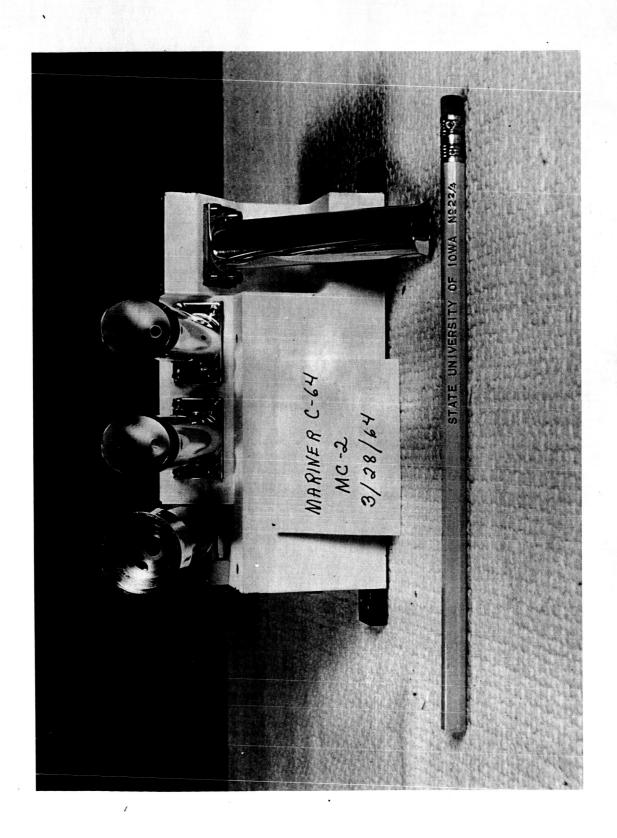


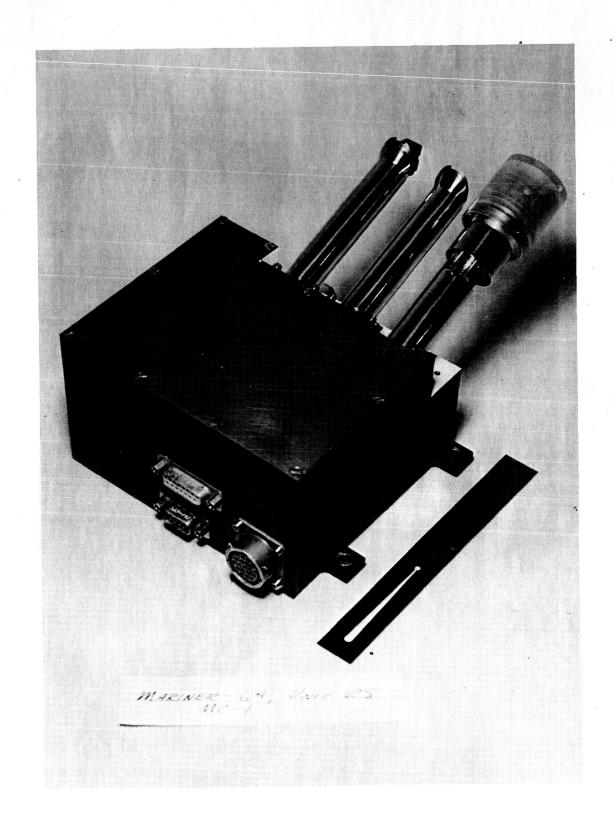


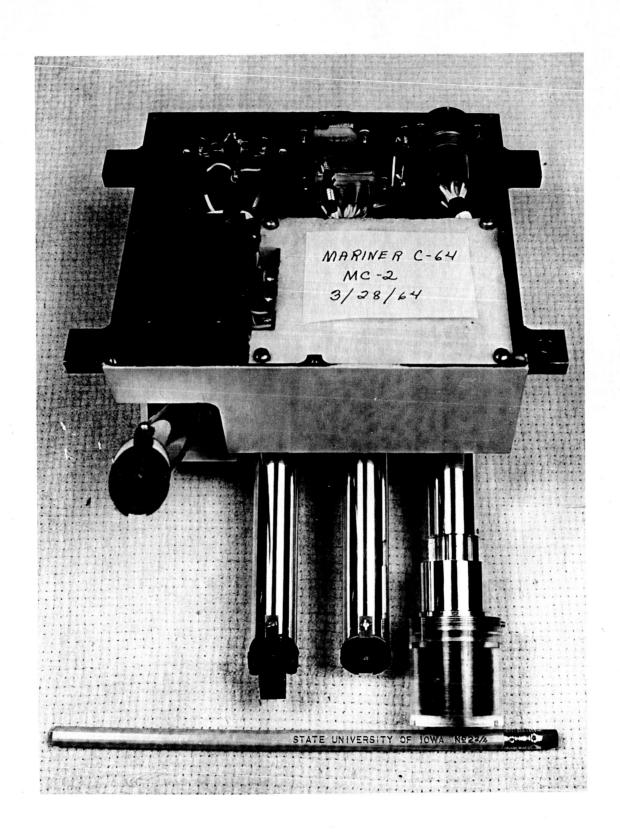
Appendix II PHOTOGRAPHS

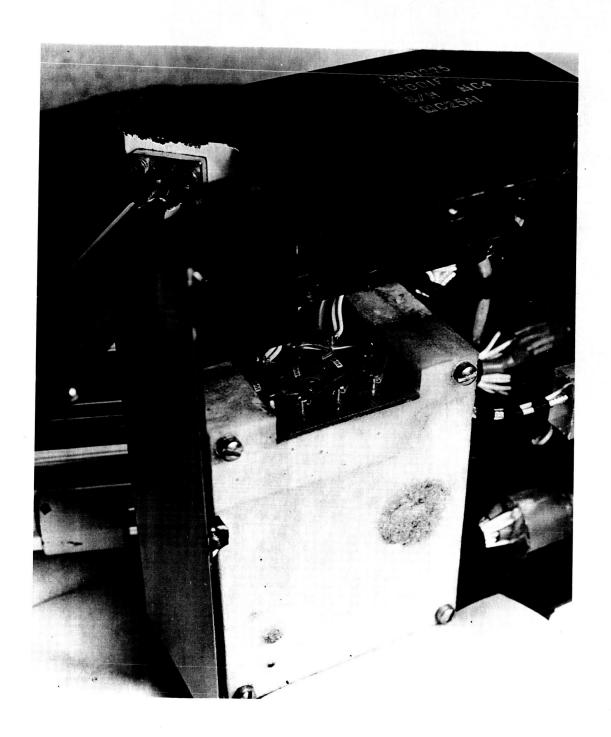


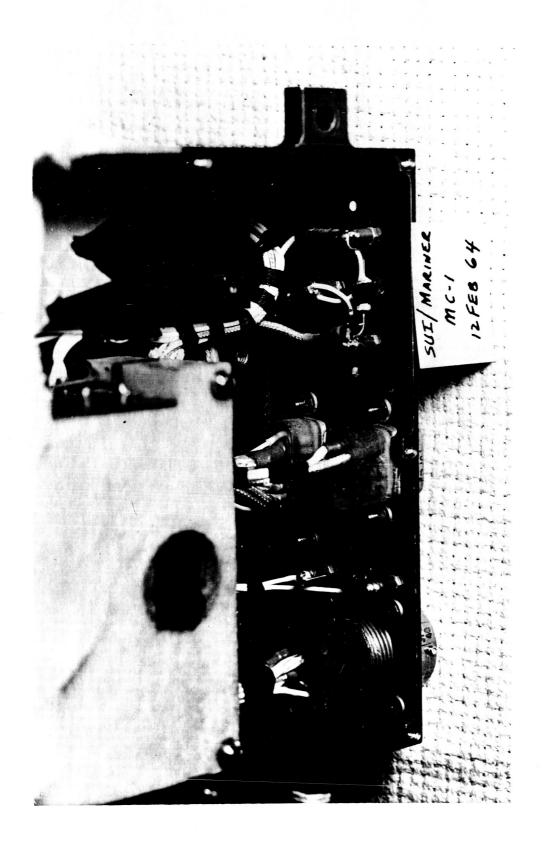


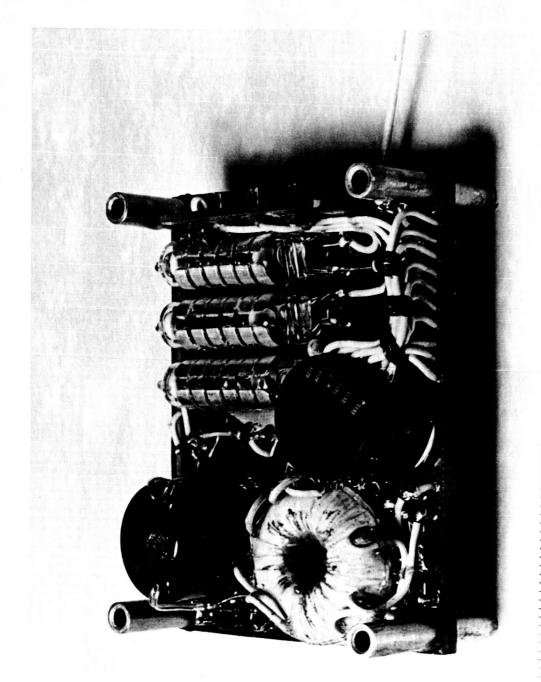




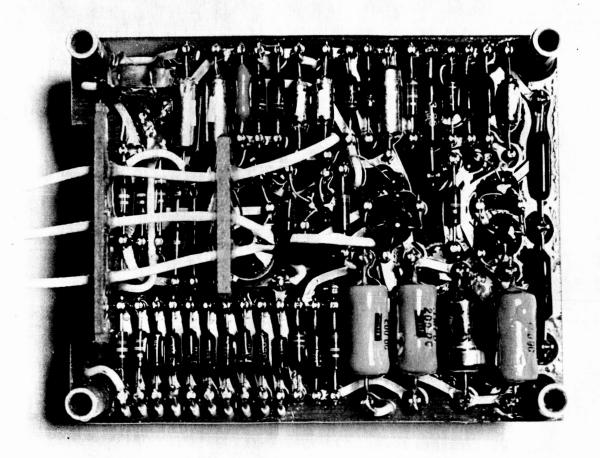




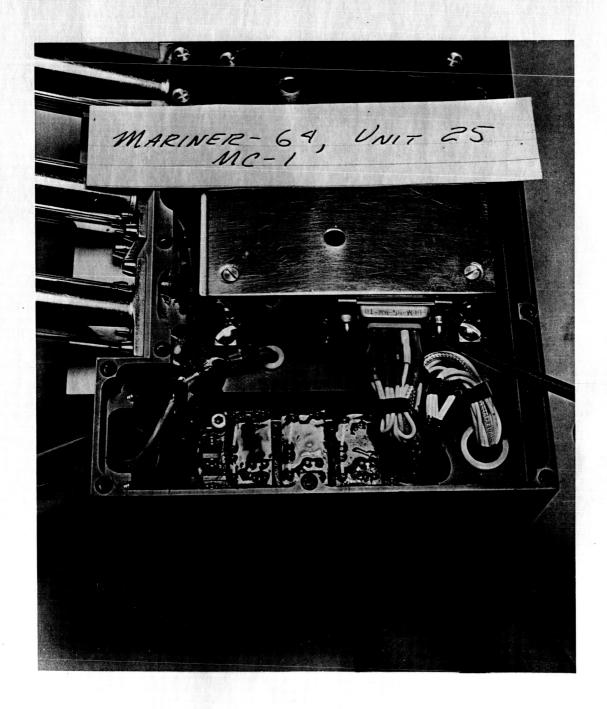


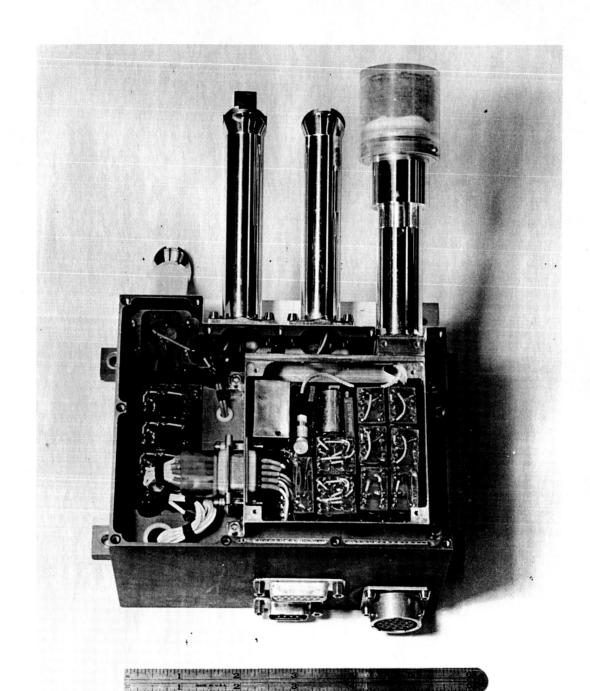


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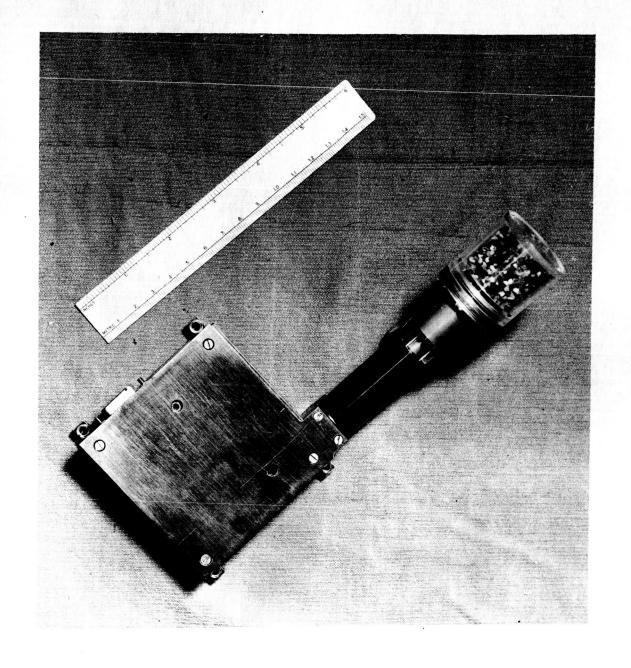
MC-1

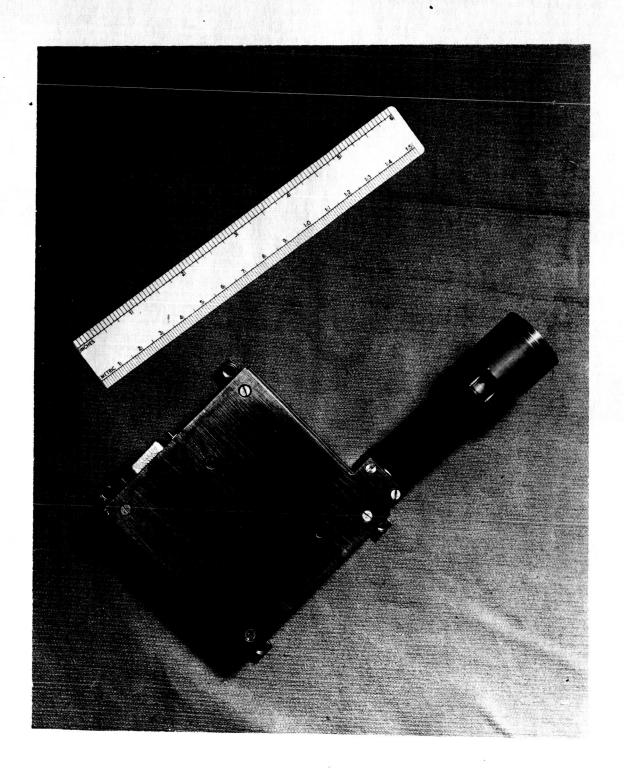


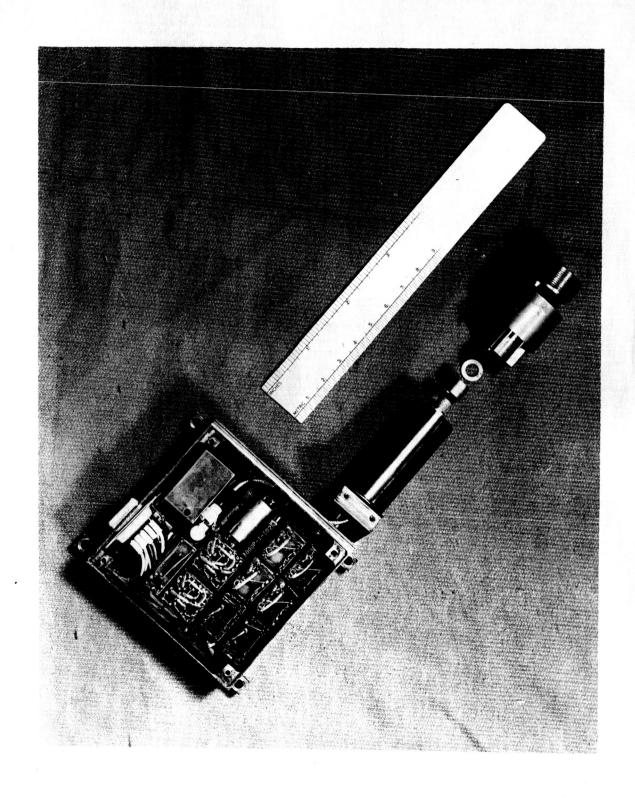


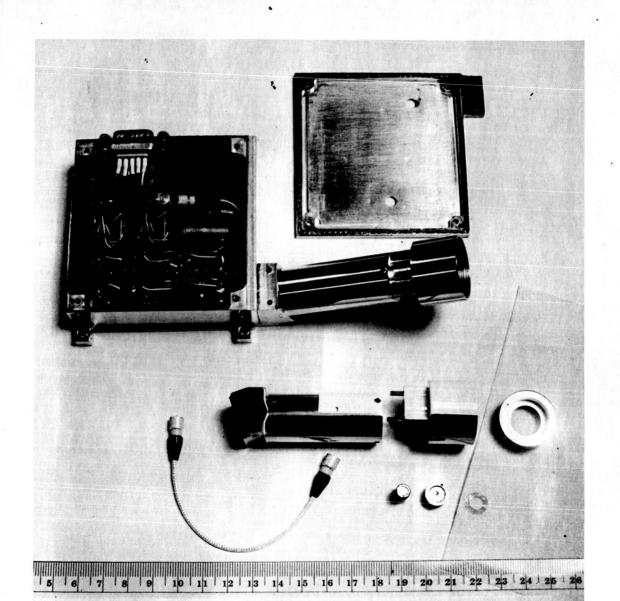
SUI /MARINER C-64 MC-1 EXPERIMENT











SUI MARINER C SURFACE BARRIER DETECTOR

